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FIRST HEATED JETTISON TEST ON THE CENTAUR STANDARD SHROUD

MEMORANDUM

Lewis Research Center Cleveland, Ohio 44135 January 1974

> FIRST HEATED JETTISON (NASA-TM-X-71501)TEST ON THE CENTAUR STANDARD SHROUD CSCL 14B (NASA) -122 p HC \$8.25 121

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National Aeronautics and Space Administration Lewis Research Center Cleveland, Ohio

ABSTRACT

The first in a planned series of heated jettison tests on the Centaur Standard Shroud was conducted at NASA Plum Brook Station's Space Power Facility on November 19, 1973. first 250-second portion of the test sequence involved heating the shroud with a specially-built fixture designed to provide a simulation of the heating environment encountered by the shroud during its ascent through the earth's atmosphere. The two heater halves, which were mounted on a rail system, were then retracted. This was followed by the jettison of the two shroud halves into catch nets positioned at 90° to the heater rails. The condition which made this test unique compared to the planned subsequent tests was the location of the maximum thermal line at 32° from the shroud separation plane. Information on the test hardware, configuration, and sequence is presented. Shroud thermal and deflection data encountered during the heating portion of the test sequence is compared with free-skin design temperatures in various graphical formats.

INTRODUCTION

The Centaur Standard Shroud protects the payload of the Titan-Centaur launch vehicle during the ascent phase of the flight. To conserve weight, it is jettisoned as early in the flight as possible, while it is still hot from aerodynamic heating. Analysis of the possible flight trajectories and the shroud structure indicated that severe internal stresses could be built up prior to jettison. Calculations of the edge motion of the shroud during jettison indicated that the design clearance between the shroud and the payload could disappear, in the worst case.

An experimental program was conducted in the Space Power Facility at the NASA Lewis Research Center to verify the computer model of the shroud jettison event. The shroud was heated to simulate the expected 280 second flight trajectory and then it was jettisoned.

A seven-megawatt, radiant heater was assembled in the vacuum chamber of the Space Power Facility. The heater was programmed to produce the desired temperature distribution with the plane of symmetry displaced 32° from the separation plane of the shroud. Following the 250-second heating cycle the heater was pulled away to allow the shroud to be jettisoned. A special catch net system was built which allowed one half of the shroud to fall completely free of the launch vehicle while the second half rotated approximately 16° before being caught. The test was performed in a 20-torr enviornment.

It is the objective of this report to present a brief description of the test hardware, the operation sequence and the results of a preliminary data analysis.

APPARATUS

The overall arrangement of the test hardware in the Space Power Facility is shown in figure 1. The seven megawatt heater was built in two halves that rolled on rails perpendicular to the facility rail system. The Centaur shroud was mounted on a Titan-Centaur interstage adapter. Its location in the 100 foot diameter chamber was chosen to allow one half of the shroud to fall completely free of the hinges before being caught in the net. The other half (the one with the dome) was caught after only 16 degrees of rotation. The Centaur tanks were not used in this test because of the unnecessary complexity they would have added. In their place, a special structure was mounted on the interstage adapter that allowed access to the inside of the shroud and supported the flight truss adapter, equipment module and a simulated pay-load. A photograph of the internal structure is shown in figure 2.

Catch Nets

Special catch-nets were constructed using a high temperature synthetic webbing supported by 6-inch-diameter aluminum pipe frames. The photograph in figure 3 shows the full-jettison catch net in position. The full-jettison net frame was supported by cables attached to 10 disk brakes (five on each side) which served to absorb the energy imparted to that shroud half. The catch system was pretested using a model of a shroud half to insure that it would function properly without damaging the shroud.

Heater

The heater was designed to duplicate, in time and temperature, the condition expected in the ascent phase of the flight. The heater contained 5910 tungsten filament lamps inside a highly polished aluminum reflector. A detailed thermal analysis (the approach used is described in ref. 1) and extensive small scale tests were performed to verify the design concepts. The heater was divided into 18 separate control zones, 11 in the cylindrical section and 7 in the biconic section, to provide the proper circumferential temperature profiles. In addition the spacing of the lamps was varied within each zone to control the vertical distribution of heat. Because the desired temperature profiles were symmetrical around the maximum heat line, the 18 control zones were further divided into mirror-image half-zones (one on each side of the plane of symmetry). The arrangement of the control and mirror image half-zones is shown in figure 4. The maximum heat line for this test was at an azimuth of 328° (32° from the shroud separation plane).

Control Systems

Each control half-zone and its mirror image was powered by a separate SCR controller. The 18 controllers were programmed individually to reproduce the expected temperature vs time curve for their respective control zones. Abort limits were established to insure that the test would not proceed if any control half-zone or mirror image half-zone deviated more than a prescribed amount from its desired temperature curve.

A PDP-8 mini computer was used to conduct the test because of critical timing of events necessary. The sequence of events for this test is presented in table 1.

INSTRUMENTATION

Thermocouples, straingages, deflectometers, and high-speed motion picture cameras were used to measure the performance of the shroud during the test. Digital data were recorded every second during the test, using an XDS 930 computer. FM analog recordings were also obtained of selected parameters. The coordinate system used to define the location of sensors on the shroud is shown in figures 5 and 6. The cylindrical section of the shroud

is a complex structure composed of a corrugated outer skin bonded to a smooth inner skin supported by circumferential "Z" rings. A sketch of the structure and a typical free skin thermocouple installation is shown in figure 7. Free skin thermocouples were located as far from structural masses as possible to provide the best possible measurement of the thermal environment. Free skin thermocouples at station 2626 in the cylindrical part of the shroud, and at station 2724 in the conic part were used to provide temperature feed back to the power controllers.

RESULTS

The heated jettison test was conducted on November 19, 1973, at an ambient pressure of 20 torr. The heater was programmed to produce the desired temperature distribution with the plane of symmetry displaced 32° from the shroud separation plane (see figure 4). The light half of the shroud (the one with the dome) was fully jettisoned and fell free of the hinges into the horizontal net. The other half was caught after only 16° of rotation.

Time histories of the control thermocouple readings are presented in figure 8 for the 18 control half-zones and the 18 mirror image half-zones. Included also on this figure are the desired temperature histories. Comparison of desired and measured temperatures shows that excellent agreement was obtained. The greatest deviation was in zone 4 in the mirror image half zone where a 17° F deviation was observed. The instantanteous power applied to the shroud varies according to the slope of the desired temperature curve. The measured power applied to zone 1 is presented as an illustration in figure 9. The initial peak in power occured because the shroud was cooler than the set point when the heating cycle started. The power increased gradually following the desired temperature curve. Very little power was needed near the end of the cycle because the required temperature was actually decreasing slowly.

Circumferential temperature profiles are represented in figure 10 at several stations and for several times during the heating cycle. Also shown are the desired temperature profiles. Comparison of the two indicated that very good agreement was obtained everywhere except at the top of the biconic section and at station 2250. The thermocouples at station 2250 are very close to the aft seal bulkhead which probably accounts for their low readings. This is supported by the fact that a thermocouple at station 2469, where the shroud skin thickness and the lamp spacing were the same as at station 2469, agreed very well with the desired curve. These deviations were observed early during the heater checkout tests and were deemed acceptable.

The shroud deflected during the heating cycle because of temperature gradients of as much as 90° F in the "Z" rings. The deflection was measured with potentiometer type deflectometers. Circumferential plots of their readings at several stations and at several times during the test are presented in figure 11; for reference purposes the desired temperature curves

are also included in figure 11. Examination of these data indicated that the shroud assumed a pinched cross section with the narrow part at the shroud separation plane. The tendency for the shroud to pinch this way is resisted by the joint between the two halves. Consequently when the shroud is separated the first motion is expected to be inward. Edge motions of the shroud were recorded by high speed motion picture cameras. At the time of this writing, the camera data had not been fully analyzed and could not be included. However, it was observed that the first motion of the shroud edges was inward (about $3\frac{1}{2}$ inches) toward the payload. In addition to the cameras some short wooden sticks mounted in foam blocks were installed to indicate invasion of the payload envelope by the shroud. First inspection indicated that the payload envelope was not invaded.

Examination of the camera data also revealed small clouds of gas, smoke, or dust escaping from the separation joints. Clouds were observed at both the 0° and the 180° separation joints near the junction of the cylindrical and biconic section (station 2680). To this date the origin and composition of these clouds have not been determined.

CONCLUSIONS

A successful heated jettison test of the Centaur Shroud was performed in the Space Power Facility on November 19, 1973. The shroud was heated to the desired thermal condition with the axis of symmetry displaced 32° from the separation plane. Initial observations indicated that the shroud's first motion was inward but that it did not invade the payload envelope.

REFERENCES

1. Hemminger, Joseph A.: Computer Simulation of Temperatures on the Centaur Standard Shroud During Heated Jettison Tests. AIAA/NASA/ASTM/IES Seventh Space Simulation Conference. NASA TM X-68273, 1973.

Table 1 SEQUENCE OF EVENTS

Verify recorder start. Start heating cycle. Verify heaters started. Color movie lights on. Color cameras on. Safe zone heaters. Start heater retract. Turn on movie lights. Check heater clear. Start cameras. Arm seal pyro. Verify seal pyro fired. Arm instrument disconnects. Verify instrument disconnects fired. Arm Super-Zip for shroud jettison. Safe all systems.	Event	Test Time
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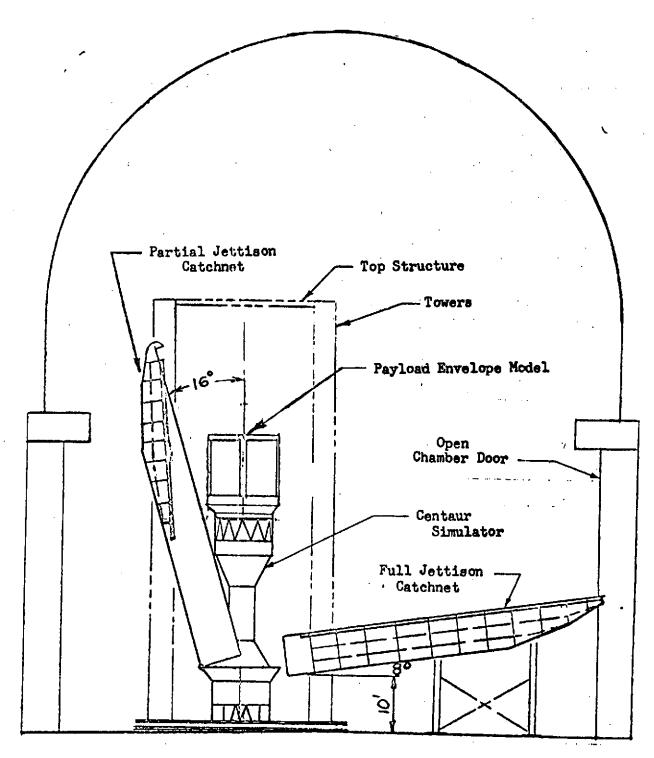


Figure 1. CSS in jettisoned position (view looking north).

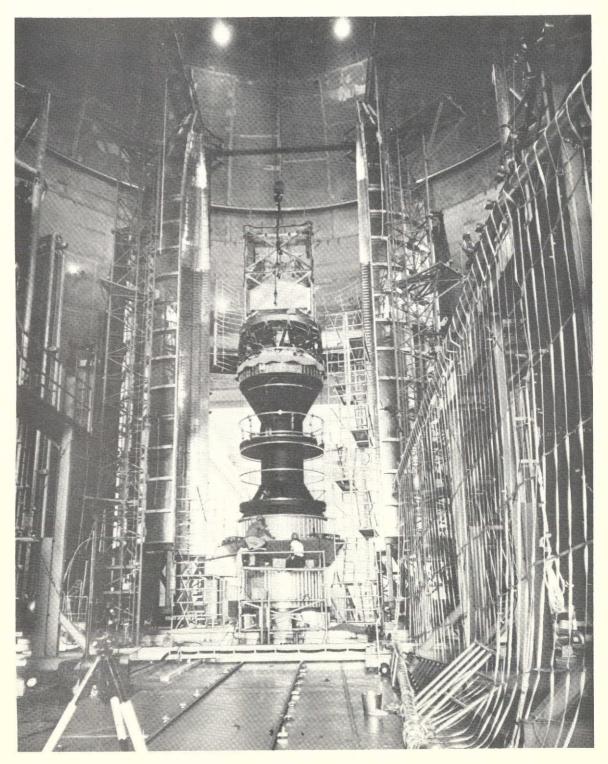


Figure 2. Photograph of internal structure.

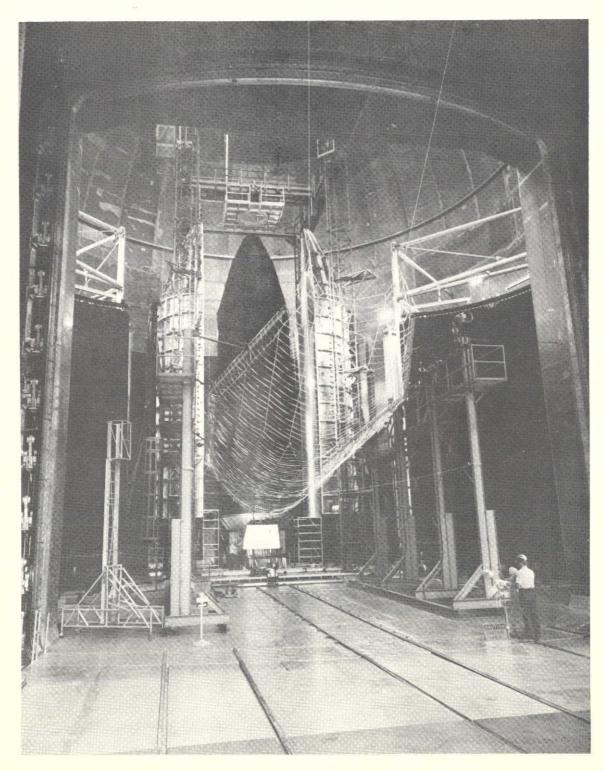


Figure 3. Photograph of test installation.

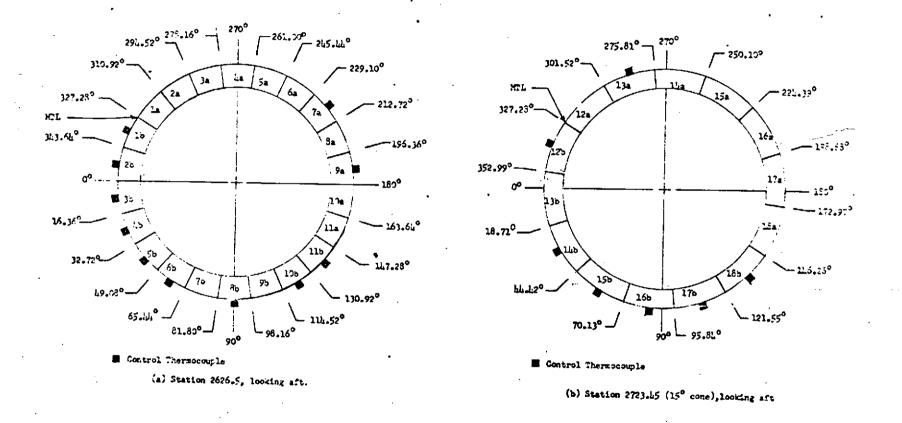


Figure 4. Heater control zones and control thermocouple locations.

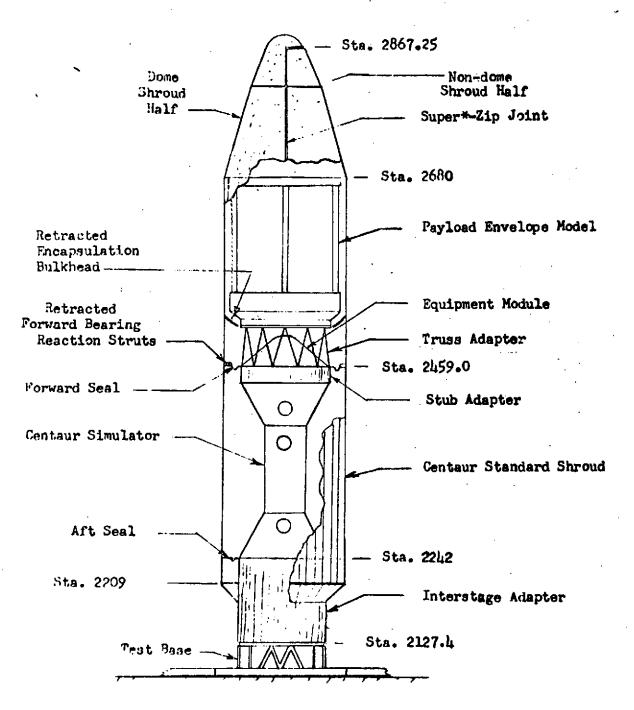


Figure 5. Axial coordinate notation - station locations (inches from datum).

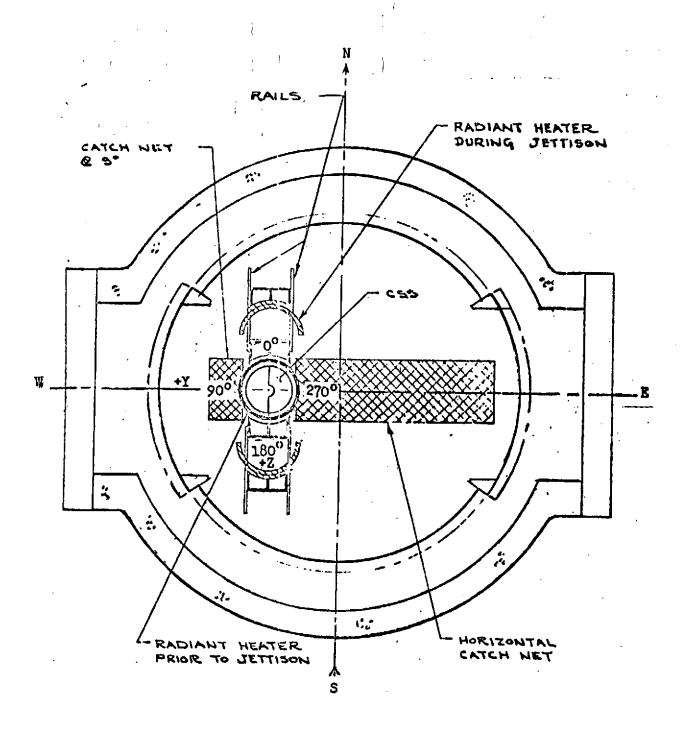
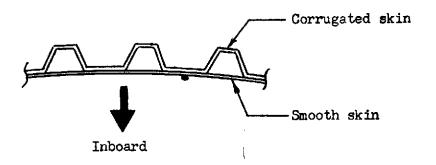
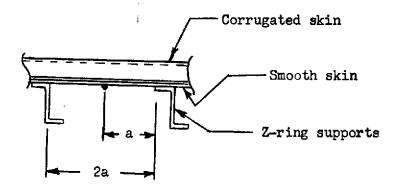


Figure 6. Angular coordinate notation - azimuth (plan view, looking down).



Note: thermocouple junctions located between "weldbonded" panels spot welds.



Typical mounting on cylindrical section

Figure 7. Free-skin thermocouple locations.

Figure 8(a) thru 8(jj). Design temperature and thermocouple histories at heating zone control locations on CSS.

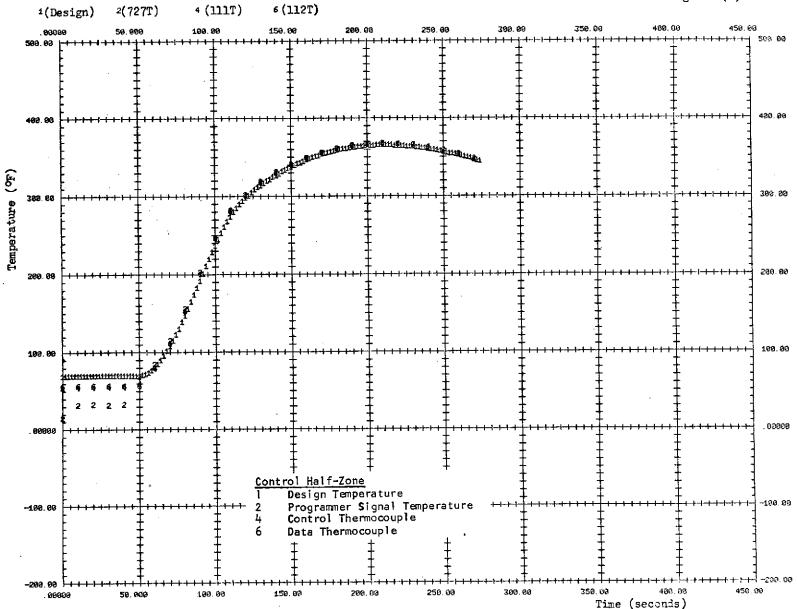
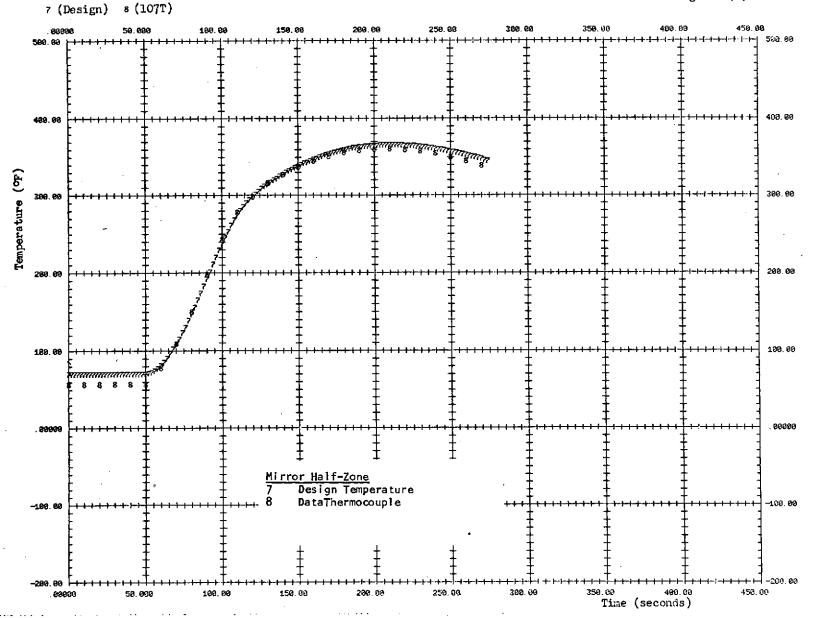


Figure 8(b)



SPF CSS-FST, RUN NO. 42 - HISTORY PLOT PLOT NUMBER 14 TIME VS. TEMP--CONT. ZONE 02 TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

9 (Design) A (728T)

c (116T)

€ (117T)

Figure 8(c)

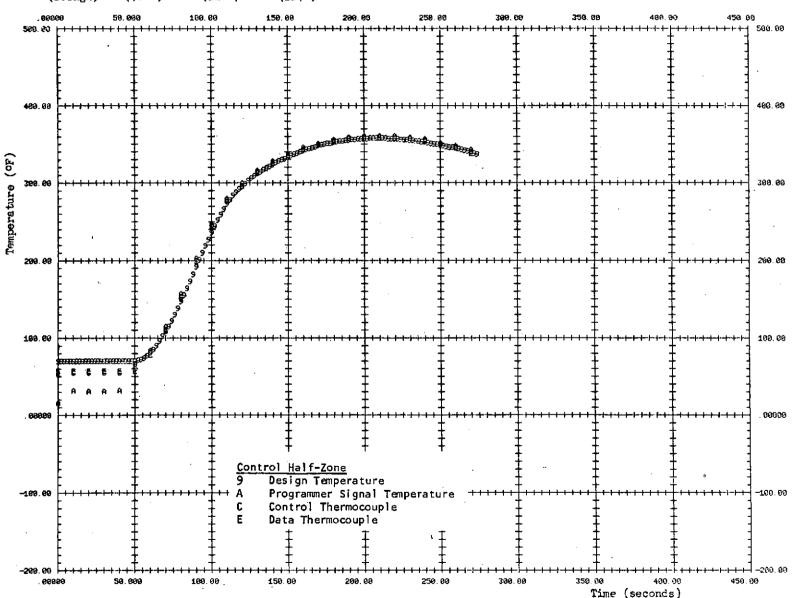
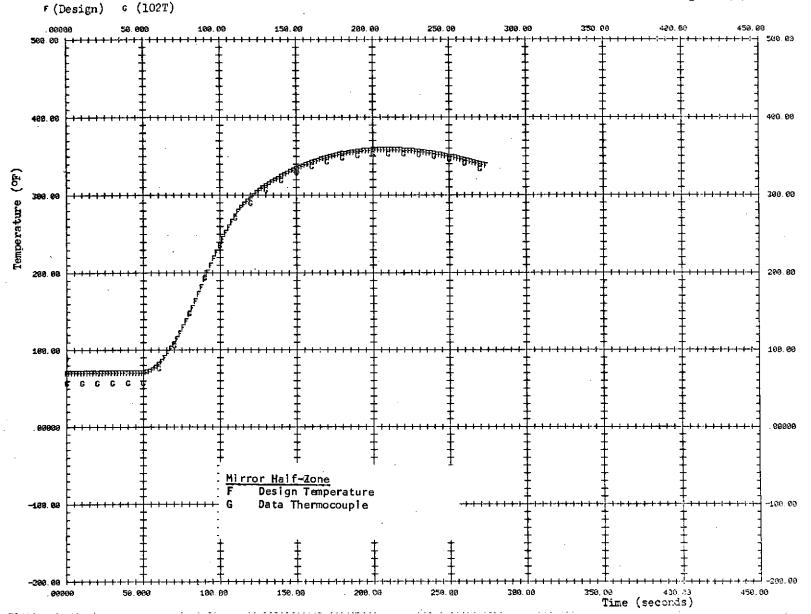


Figure 8(d)



-200.00 ----200.00

50.900

169.89

150.00

200.00

250.00

300.00

359.00

400.00

Time (seconds)

459.00

Figure 8(f)

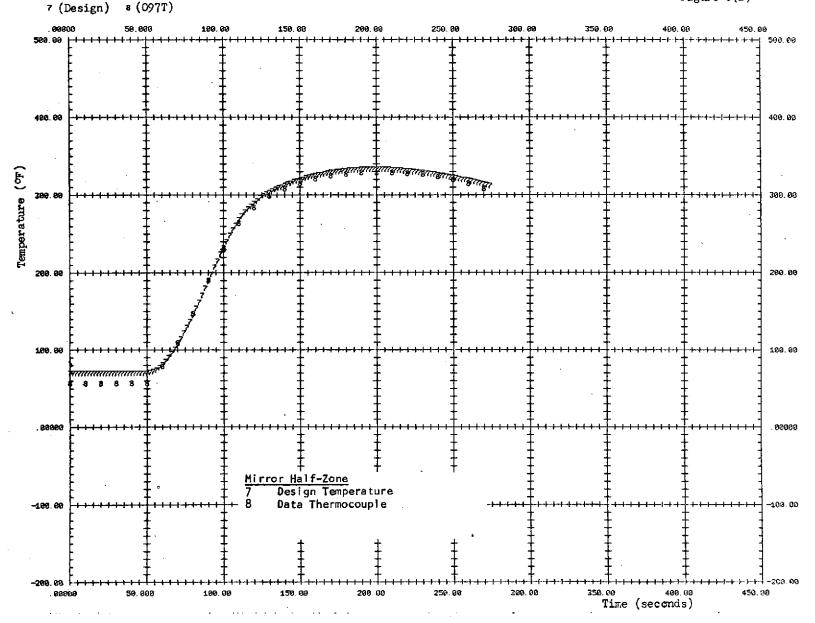


Figure 8(g)

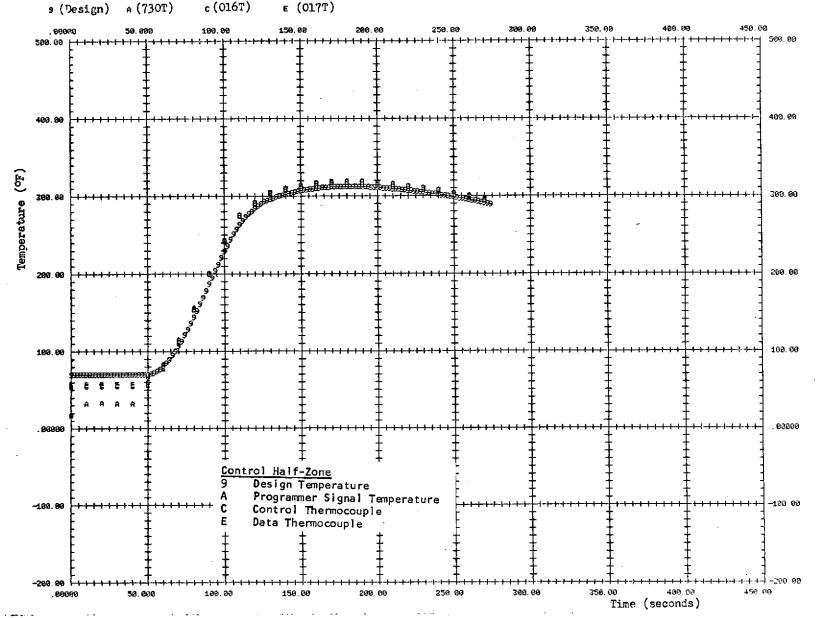
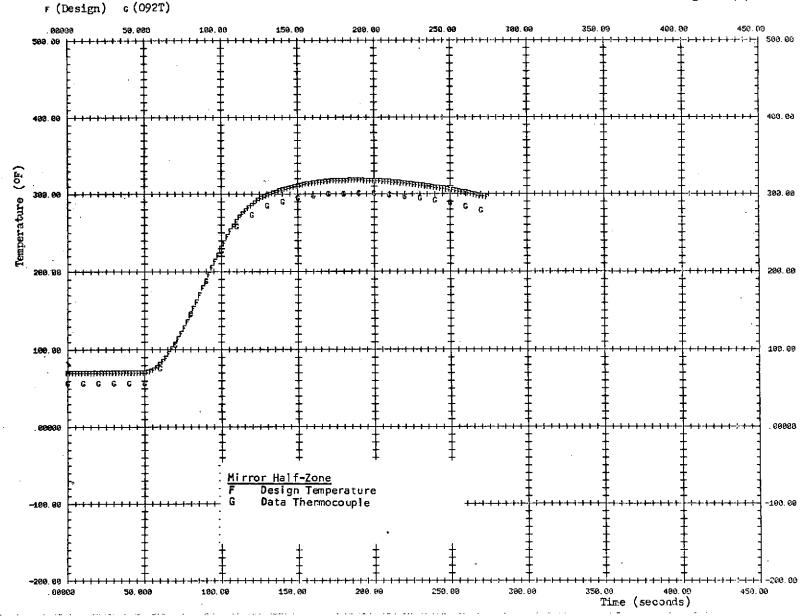
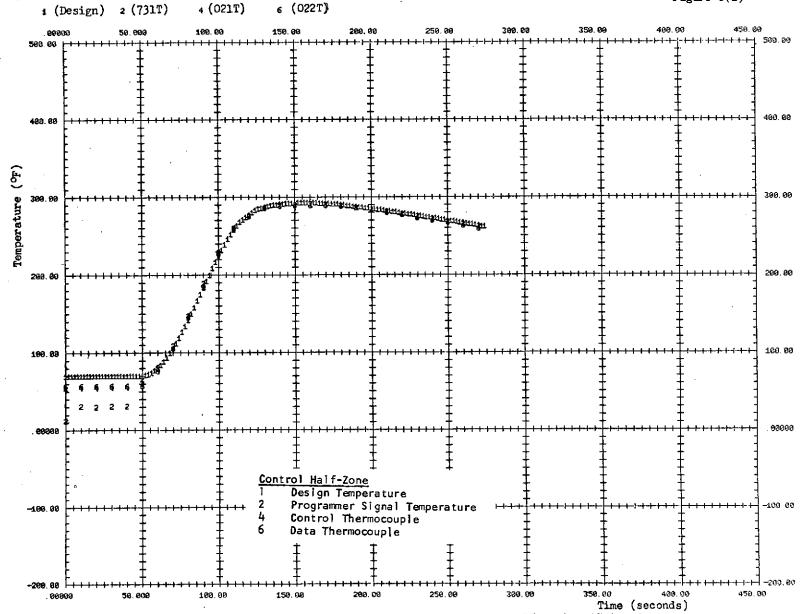


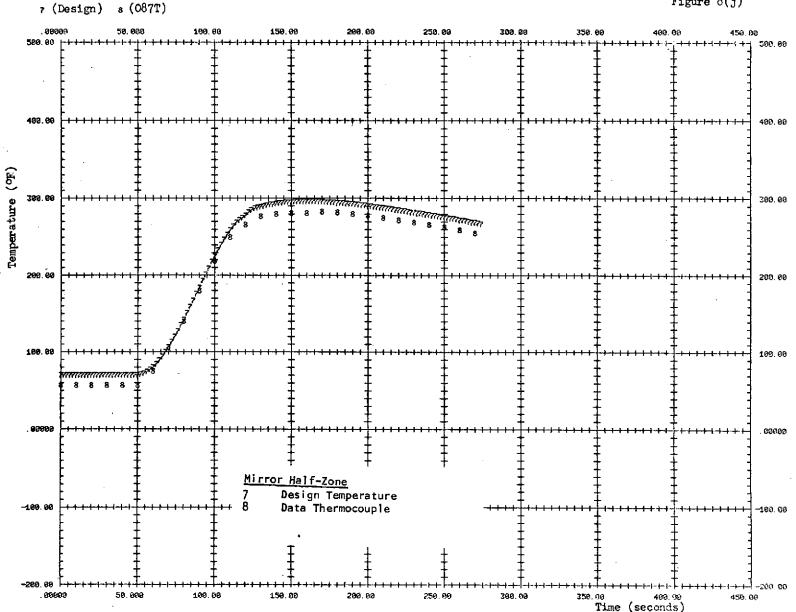
Figure 8(h)

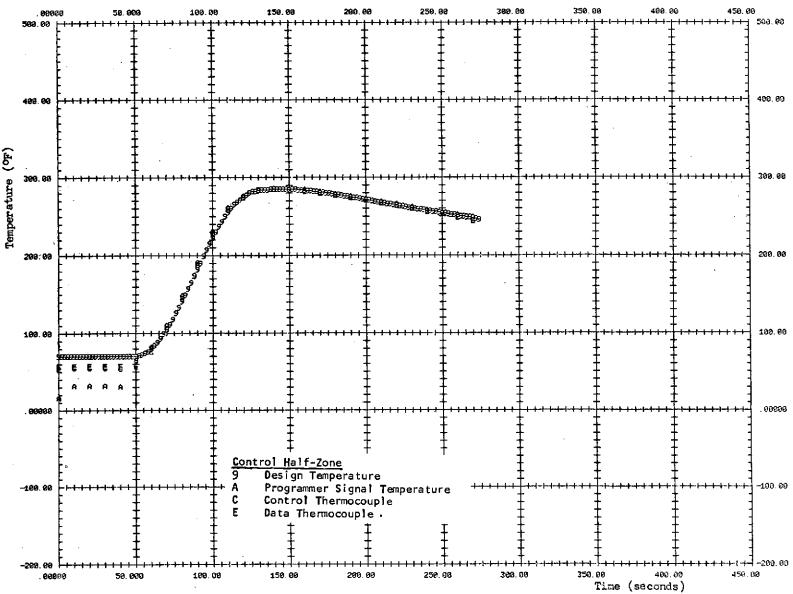


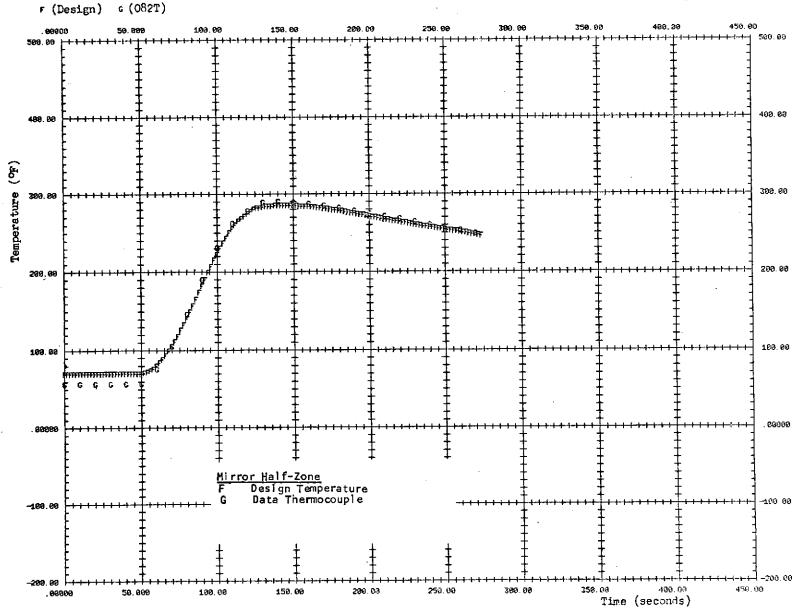
SPF CSS-FST, RUN NO. 42 - HISTORY PLOT PLOT NUMBER 06 TIME VS. TEMP-CONT. ZONE 05 TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 8(i)



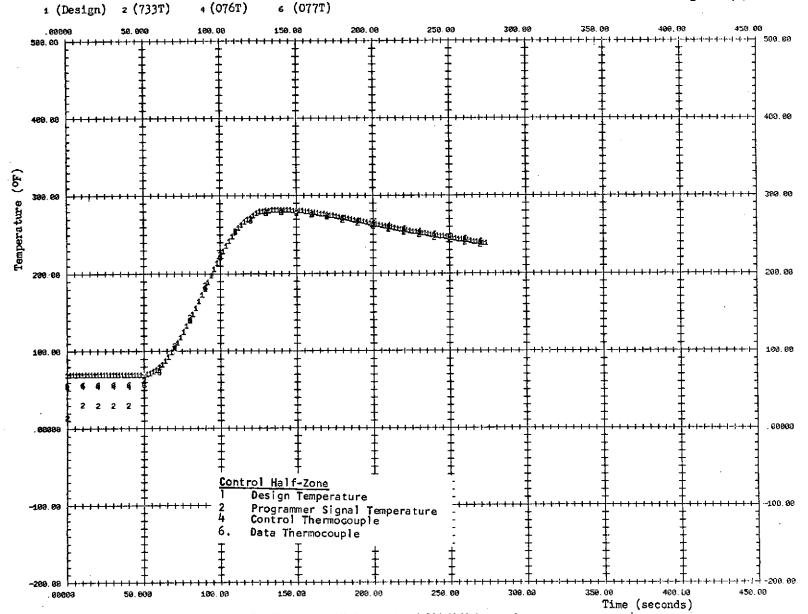


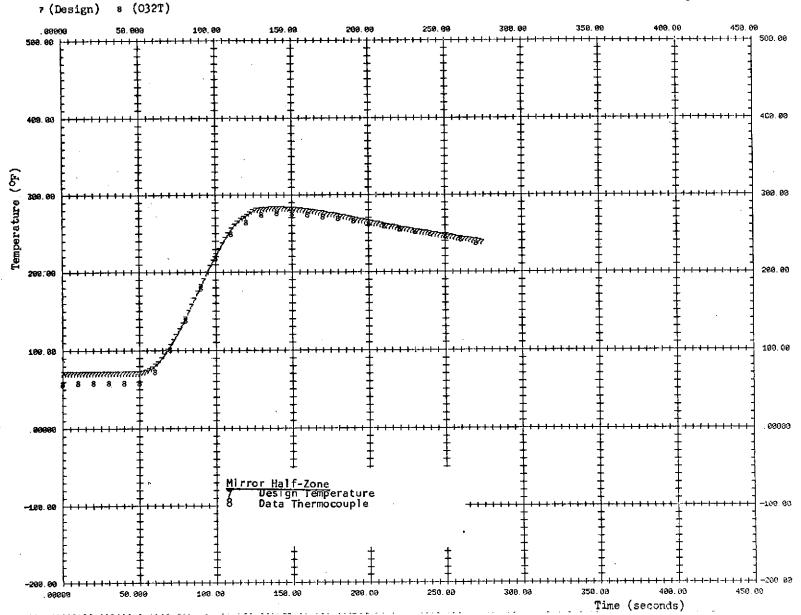




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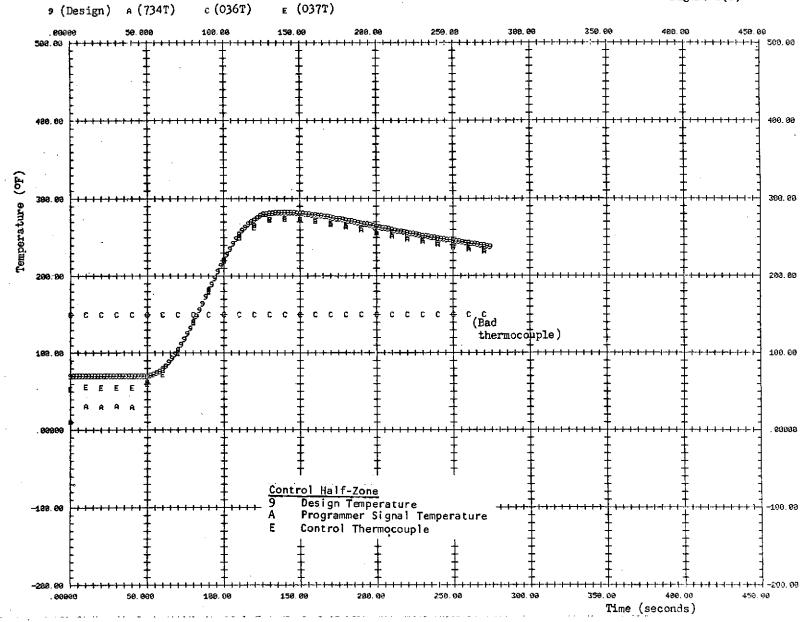
Figure 8(m)

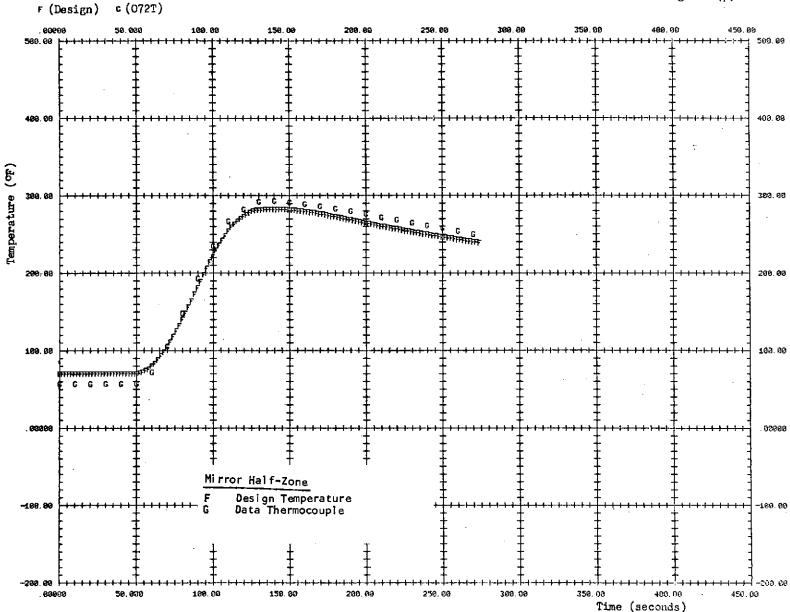




TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 8(o)





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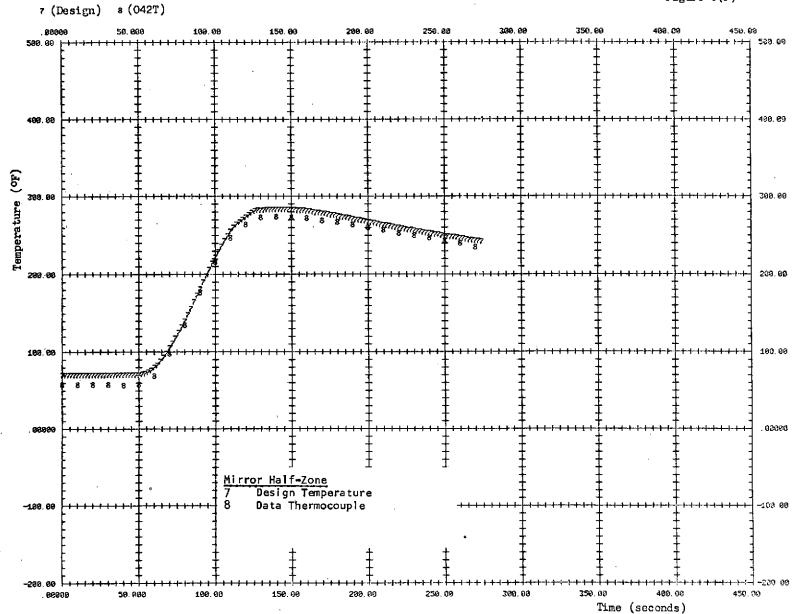
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Time (seconds)



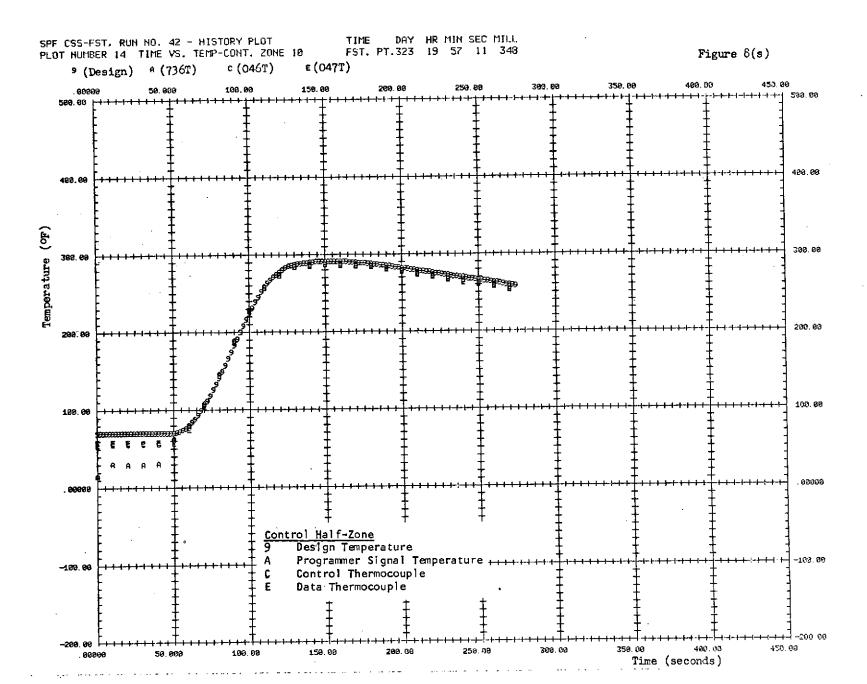
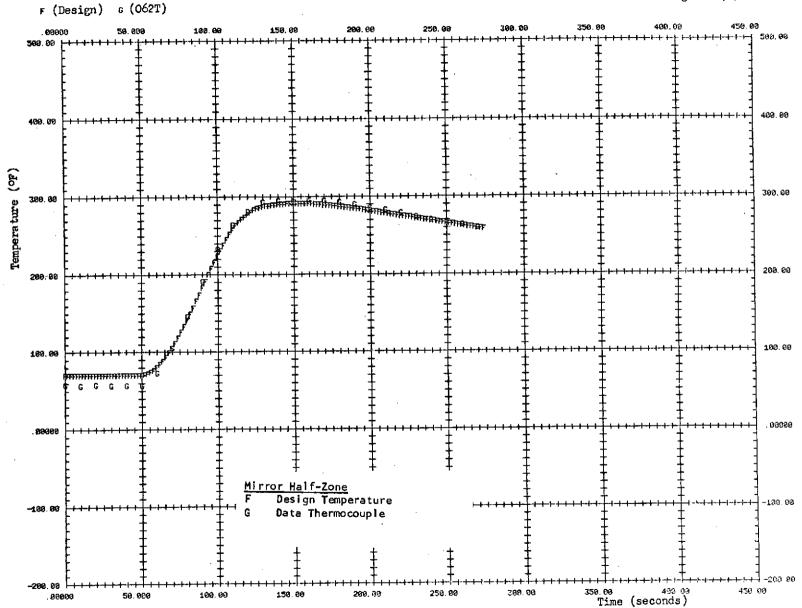
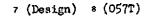


Figure 8(t)





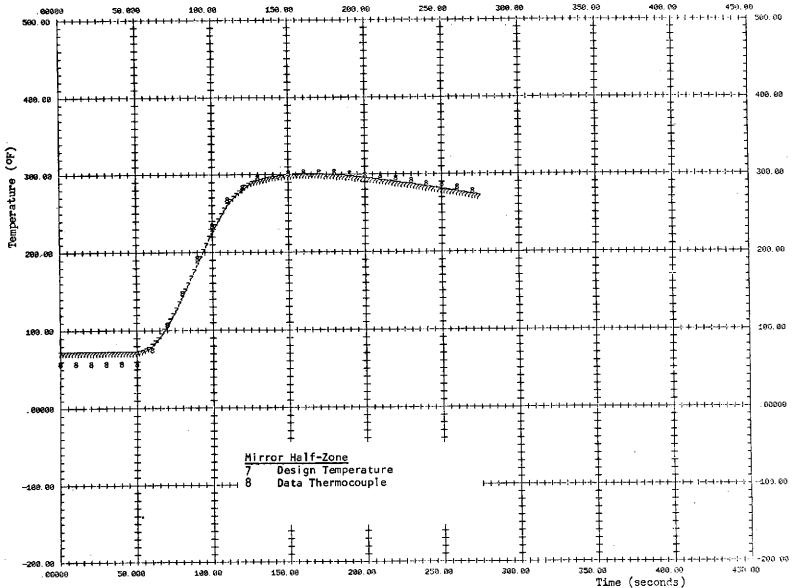


Figure 8(w)

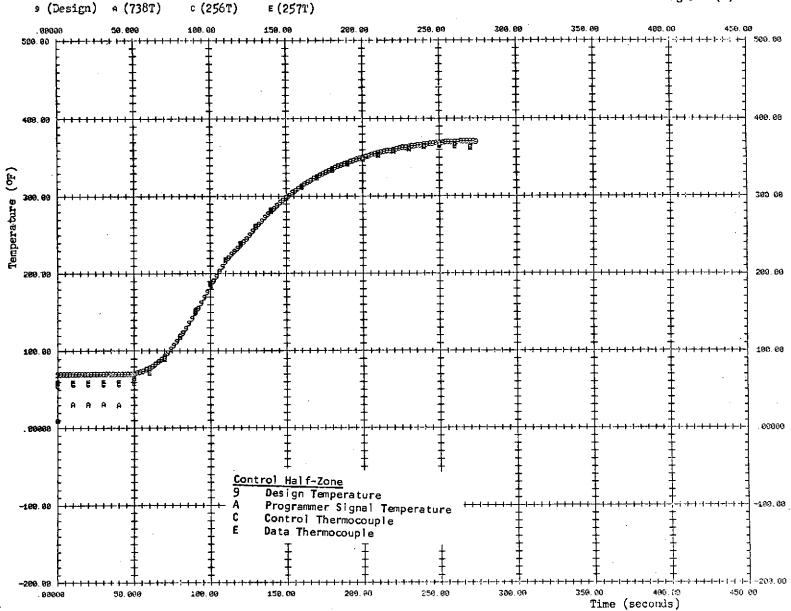


Figure 8(x)

F (Design) G (252T)

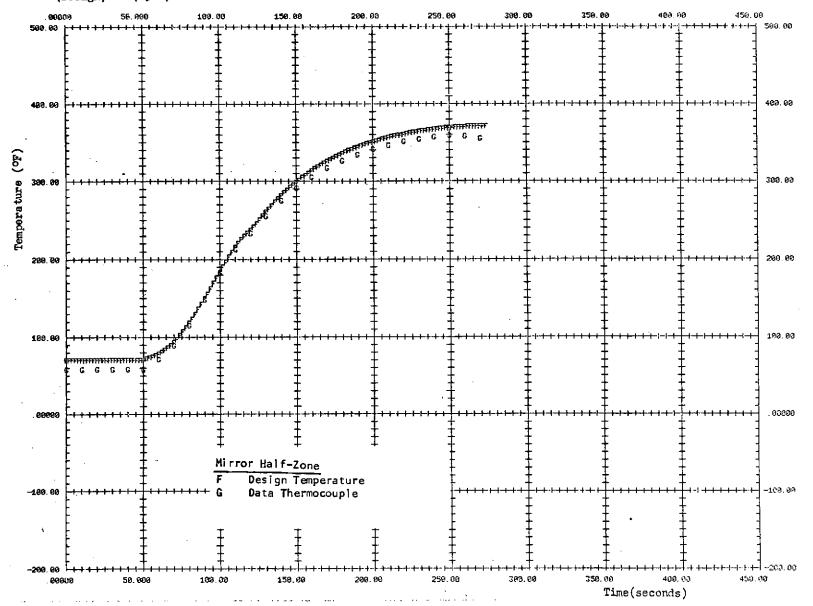


Figure 8(y)

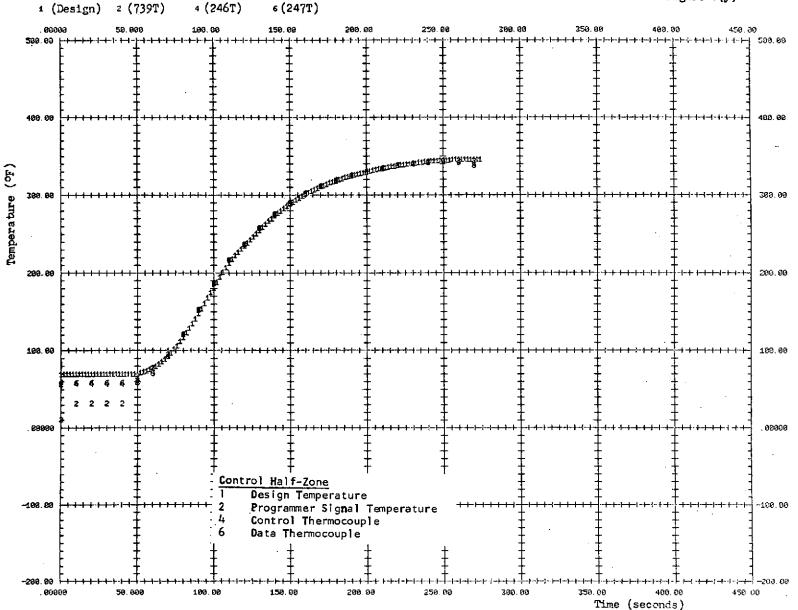


Figure 8(z)

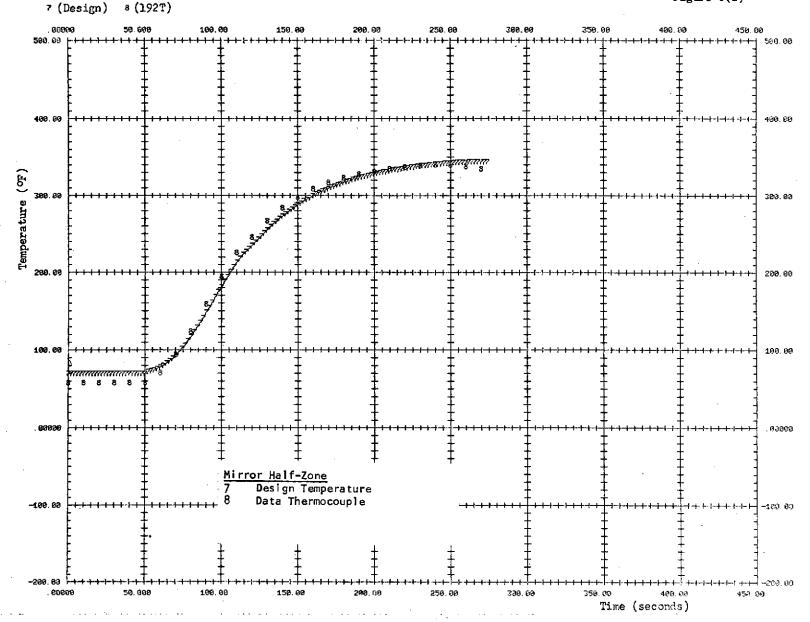
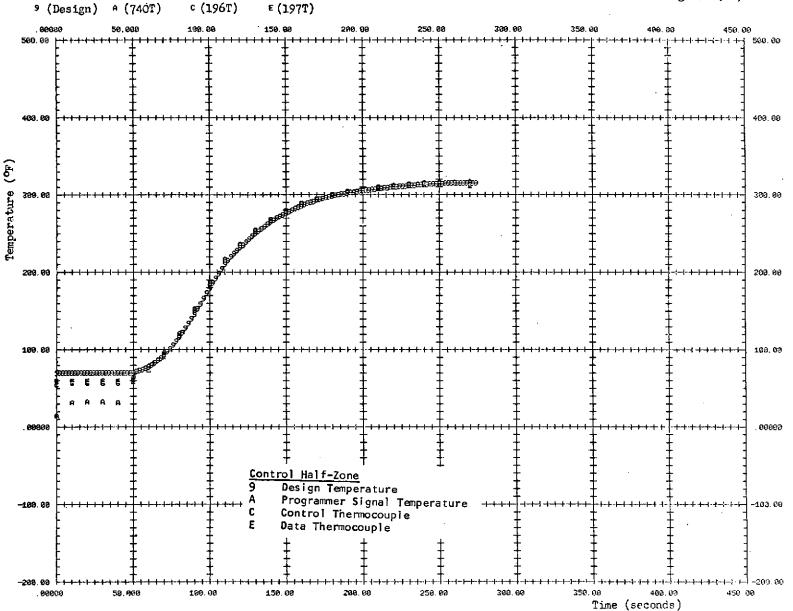
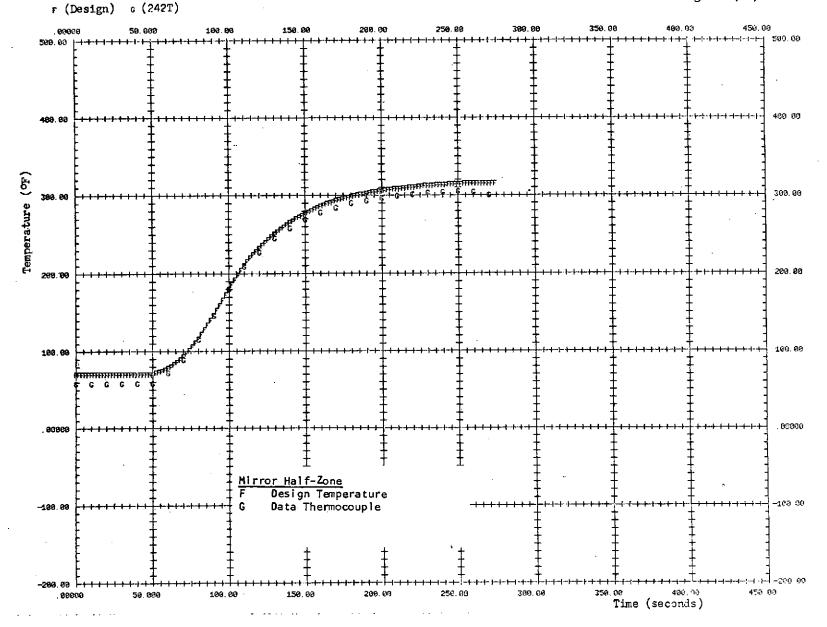


Figure 8(aa)





SPF CSS-FST, RUN NO. 42 - HISTORY PLOT PLOT NUMBER 06 TIME VS. TEMP-CONT. ZONE 15

DAY HR MIN SEC MILL FST, PT.323 19 57 11 348

6 (202T)

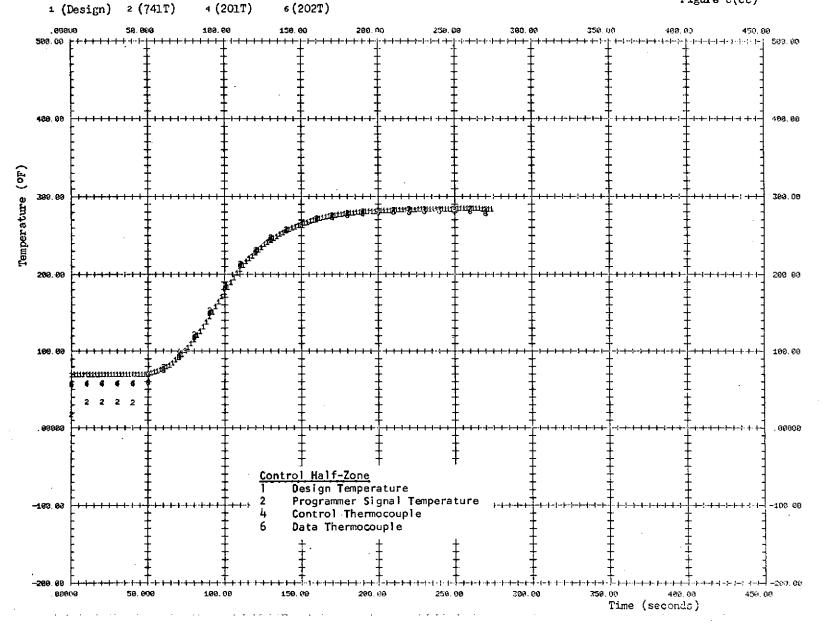


Figure 8(cc)

Figure 8(dd)

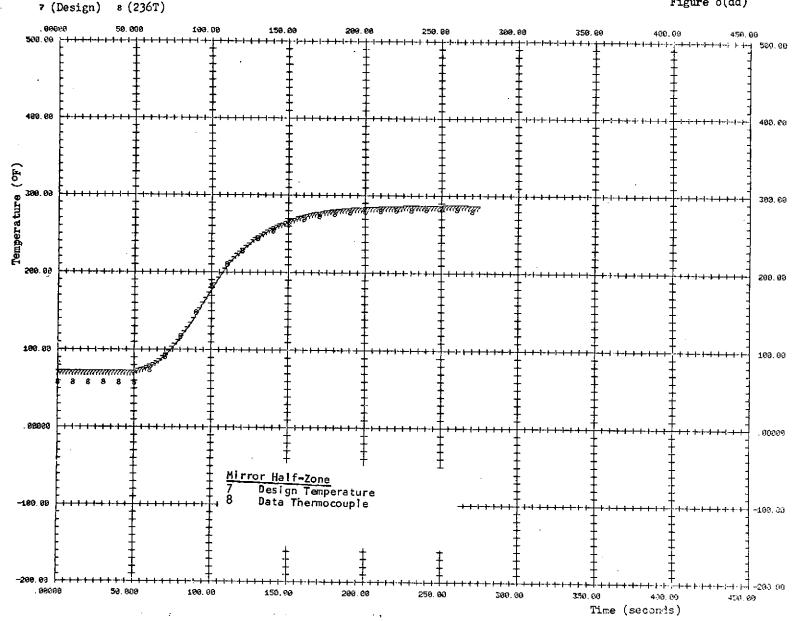
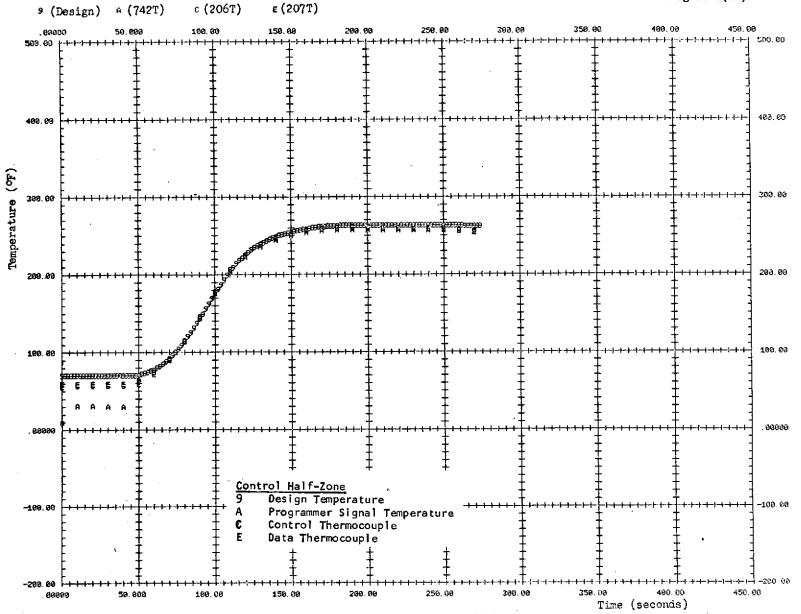
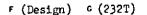
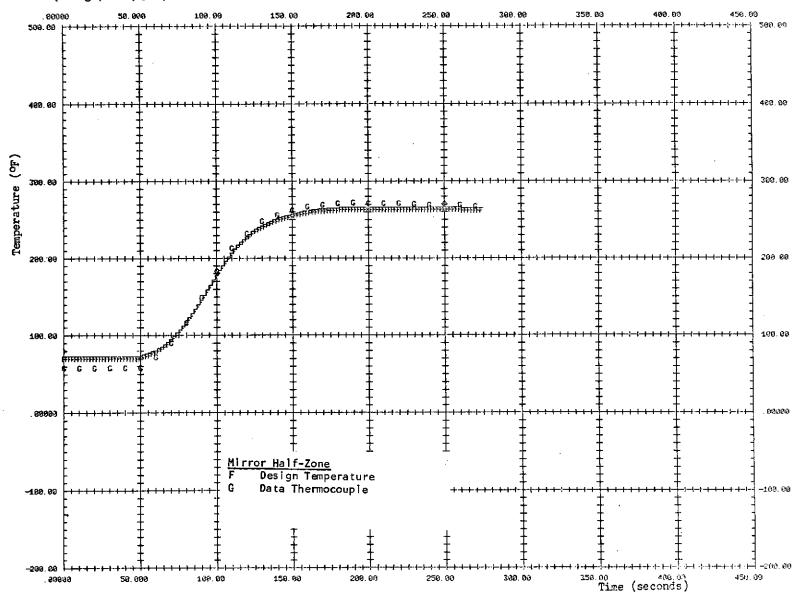


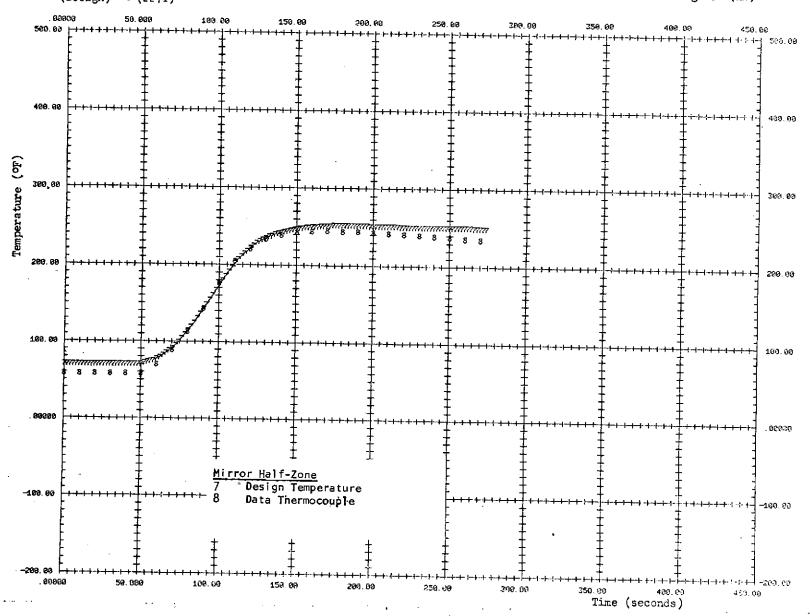
Figure 8(ee)







SPF CSS-FST, RUN NO. 42 - HISTORY PLOT DAY HR MIN SEC MILL PLOT NUMBER 06 TIME VS. TEMP-CONT. ZONE 17 FST. PT.323 19 57 11 348 Figure 8(gg) 6 (212T) 1 (Design) 2 (743T) 4 (211T) 300 00 350,60 400.00 150.00 200.00 250.00 . 69000 Temperature (OF) 300.00 นามาการแกรมการการ 4 4 4 6 2 2 2 2 . 20000 Control Half-Zone Design Temperature Programmer Signal Temperature Control Thermocouple Data Thermocouple 259.89 . 350, 00 408.00 459.00 150.00 200.00 388,99 100.00 . 60600 50.000 Time (seconds)



SPF CSS-FST, RUN NO. 42 - HISTORY PLOT PLOT NUMBER 14 TIME VS. TEMP-CONT. ZONE 18 TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 8(ii)

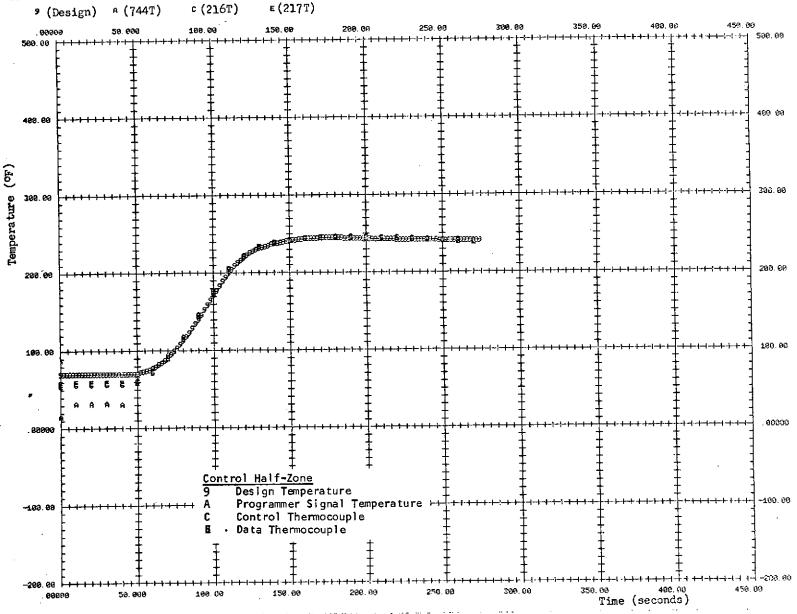


Figure 8(jj)

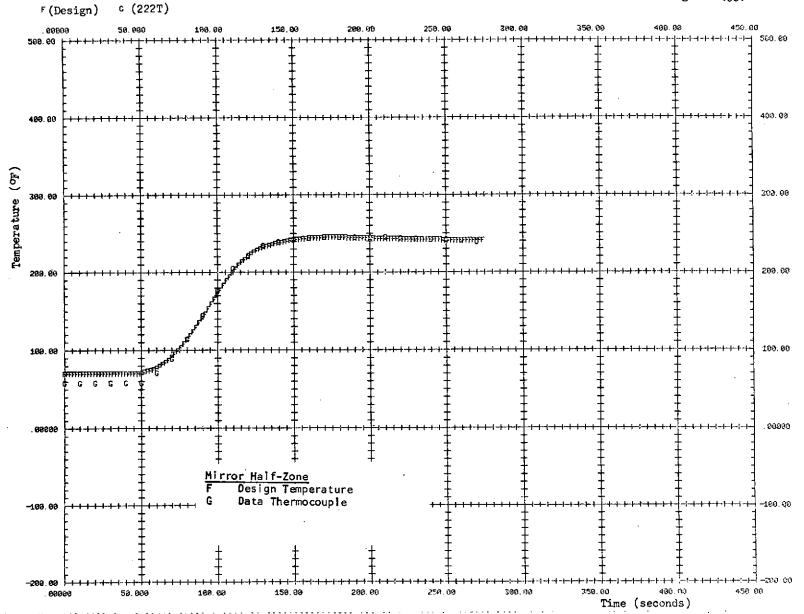
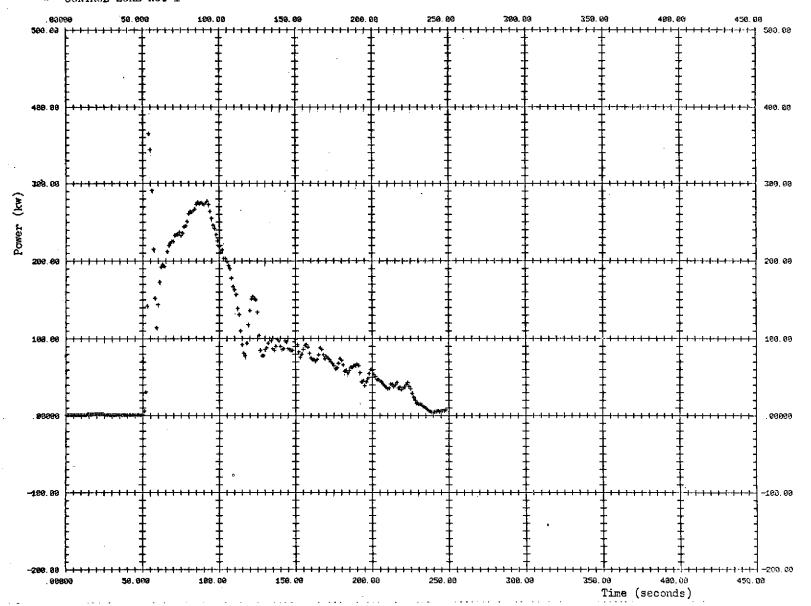


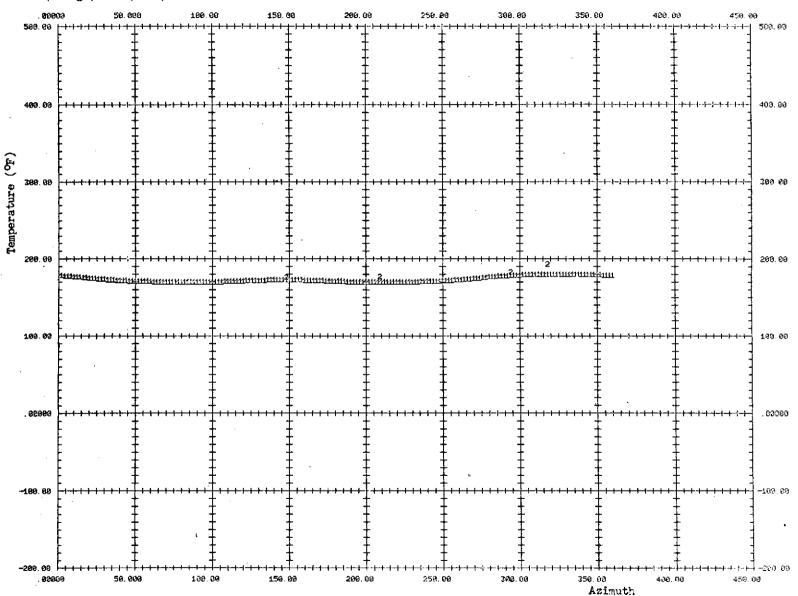
Figure 9. Power history for heating zone no. 1 power controller.

d CONTROL ZONE NO. 1



Figures 10(a) thru 10(nn). Design and data thermocouples circumferential temperature distributions.

1 (Design) 2 (Data)



SPF CSS-FST, RUN NO. 42 - CIRC. DIST. PLOT PLOT NUMBER 02 STA. NO. 2220.0, TIME 150.0

TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

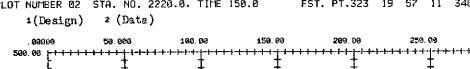
300.00

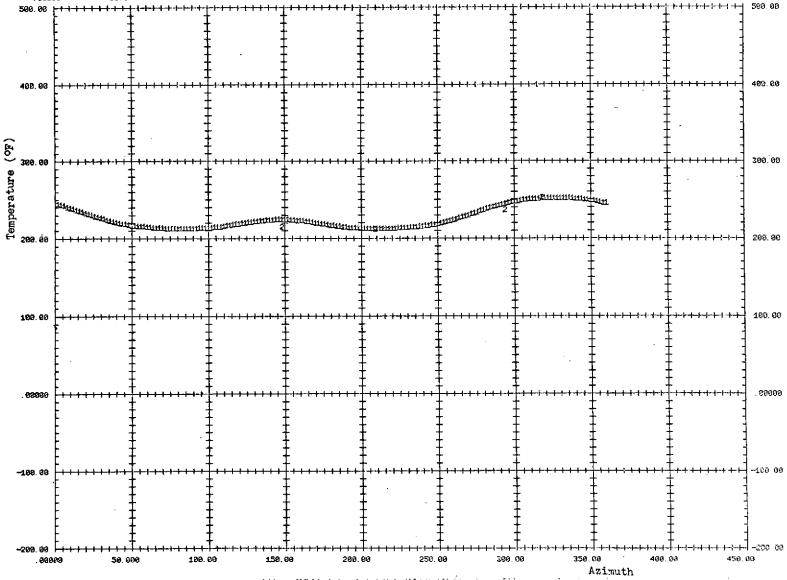
350.00

Figure 10(b)

450.00

400.00





SPF CSS-FST, RUN NO. 42 - CIRC. DIST. PLOT PLOT NUMBER 02 STA. NO. 2220.0, TIME 200.0

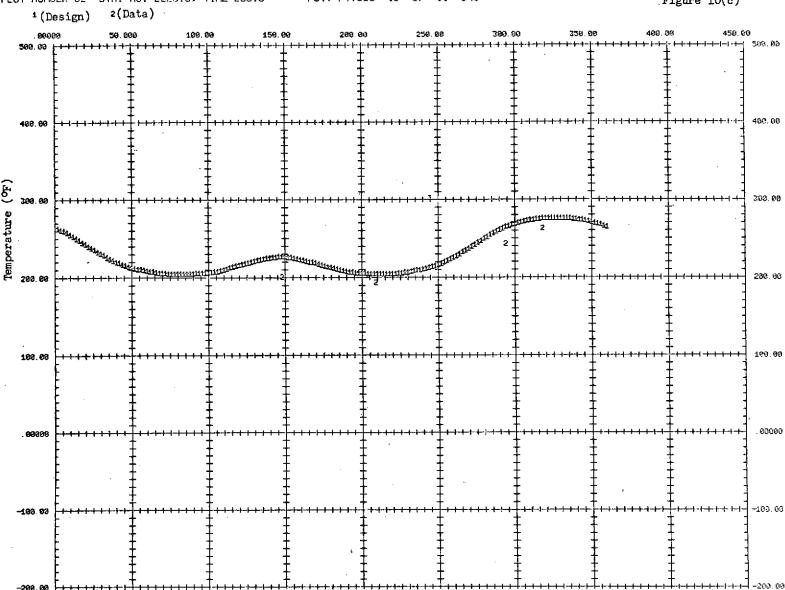
50.000

. 00000

160.00

TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 10(c)



200.00

150.00

250.00

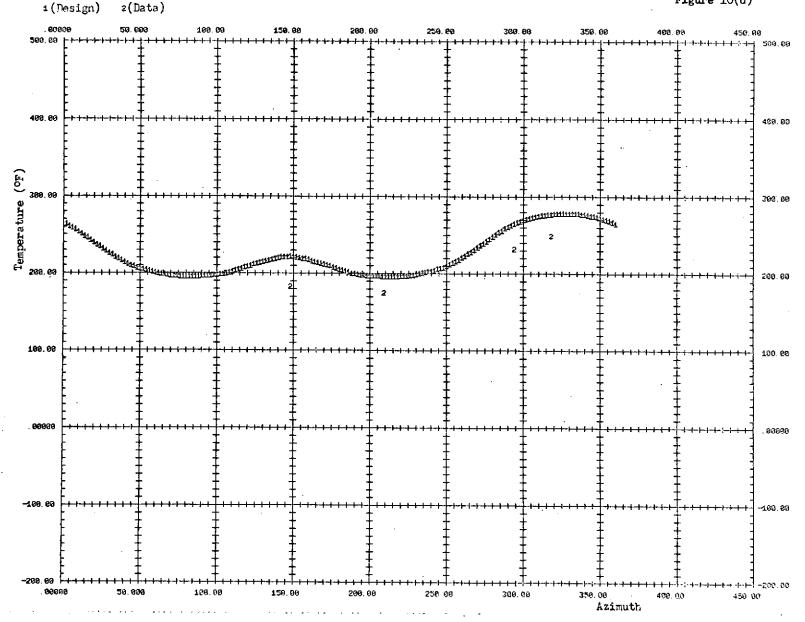
300.00

350.00

Azimuth

450, 00

Figure 10(d)



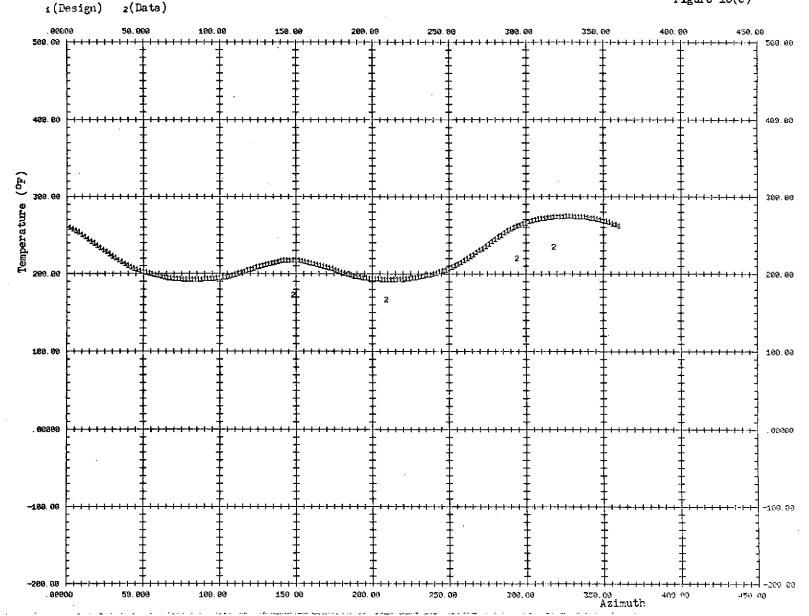


Figure 10(f)

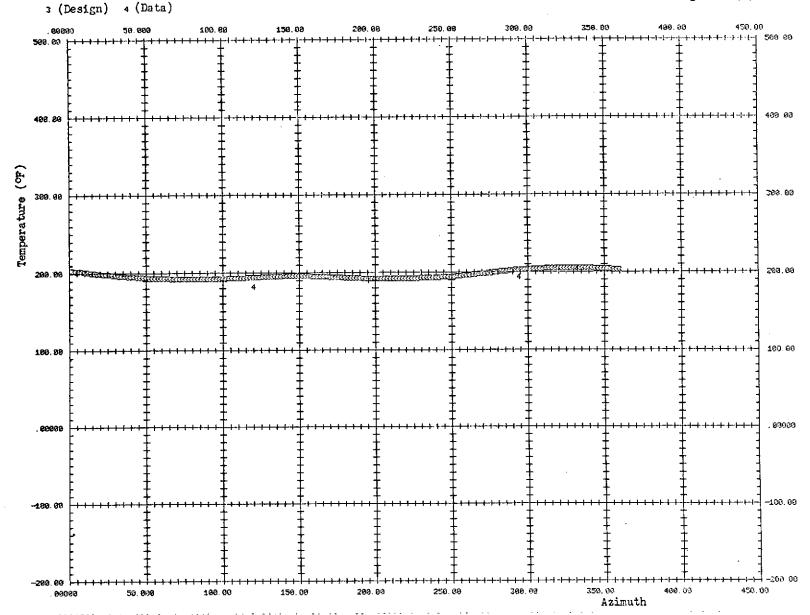
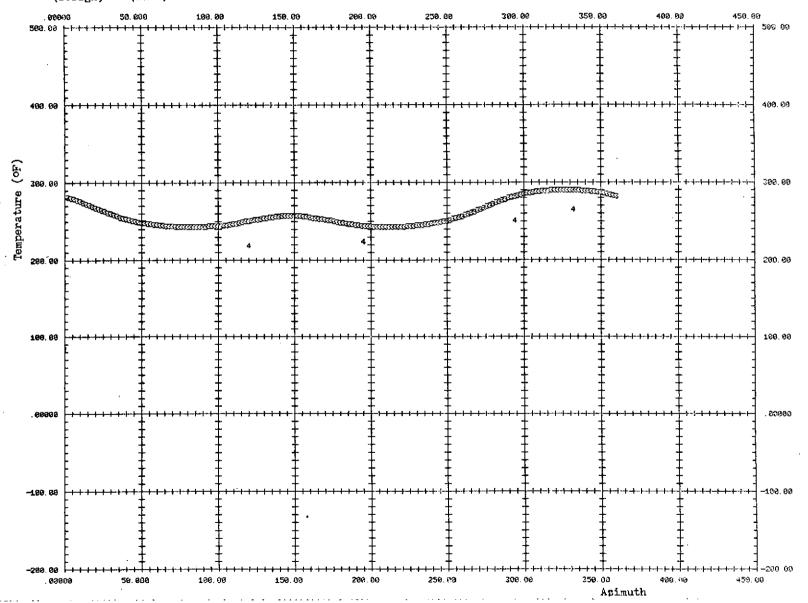
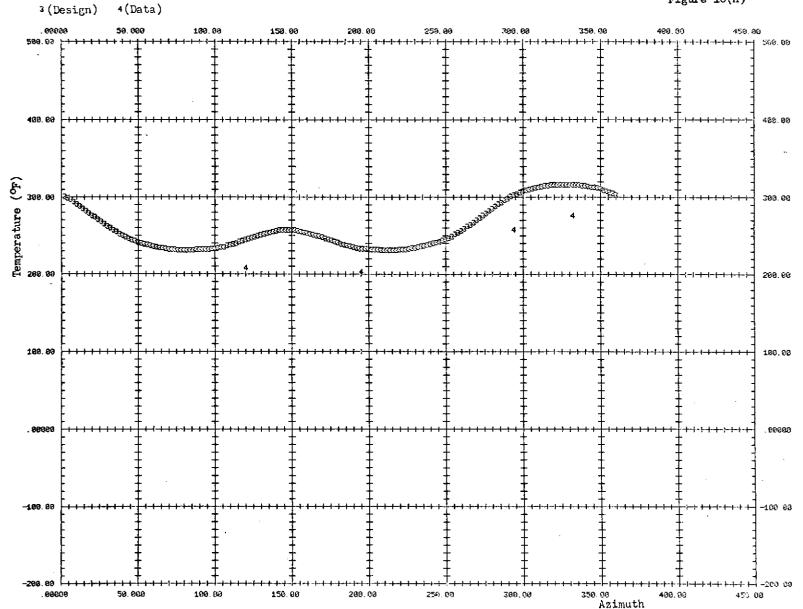


Figure 10(g)

3 (Design) 4 (Data)



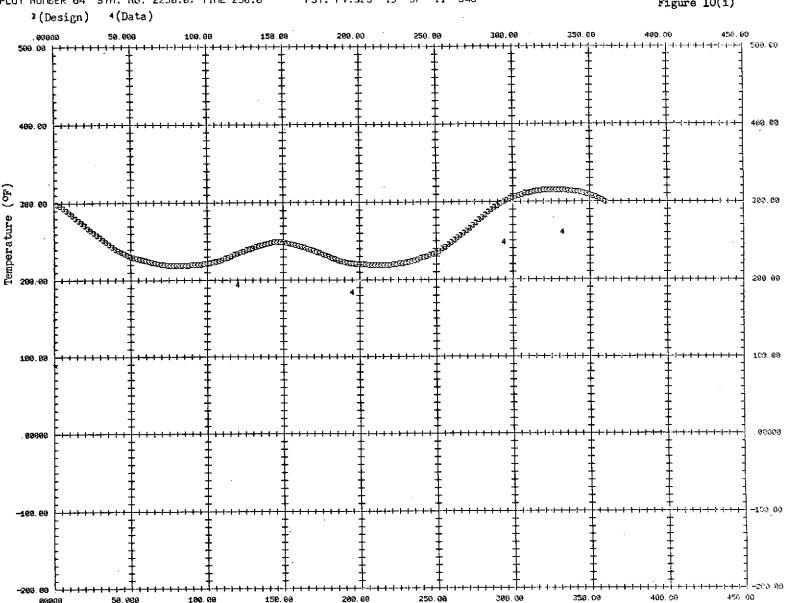


50.980

100.00

Figure 10(i)

Azimuth



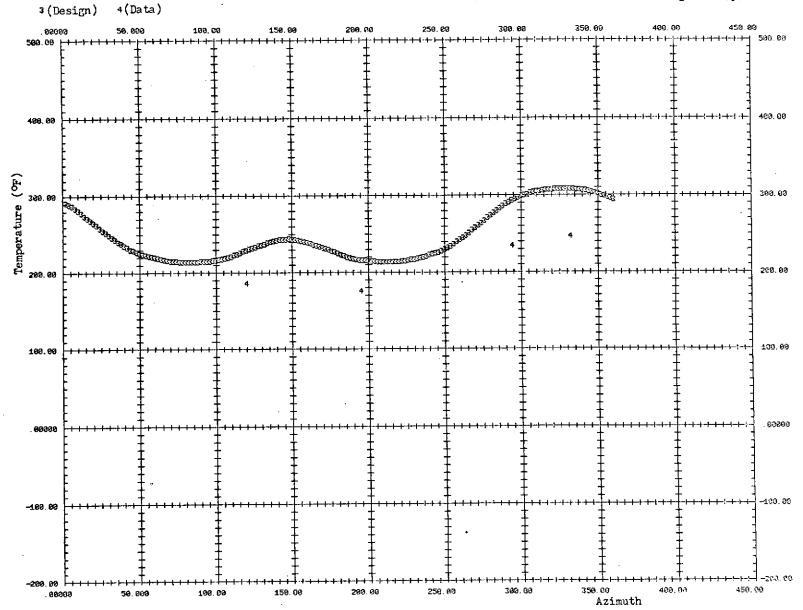
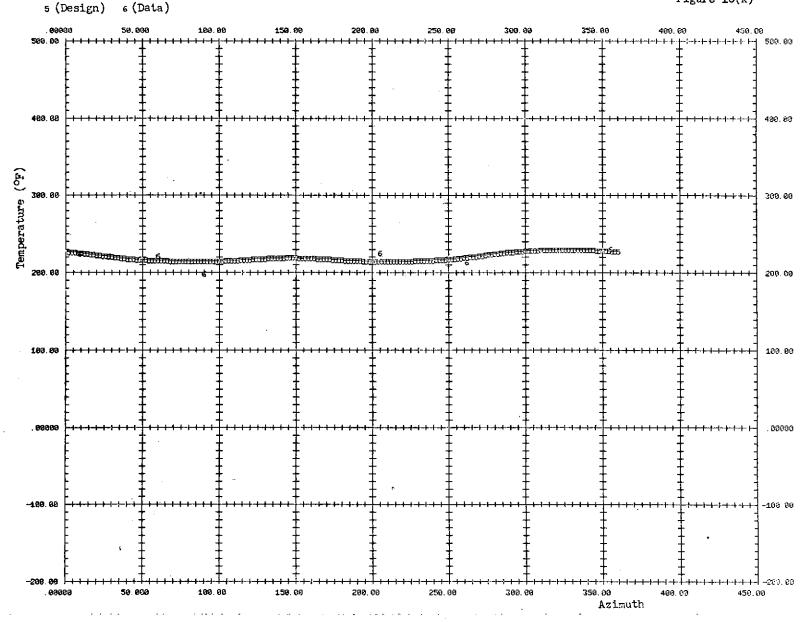
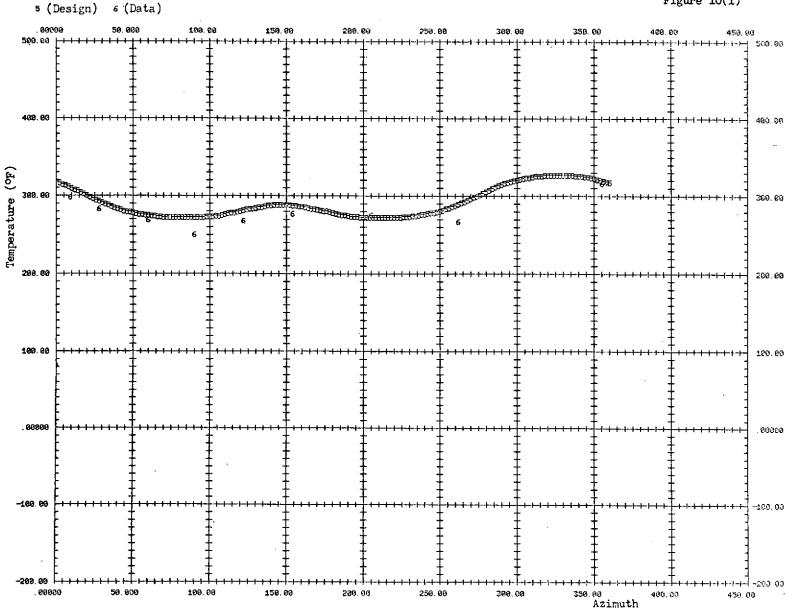
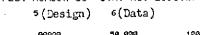
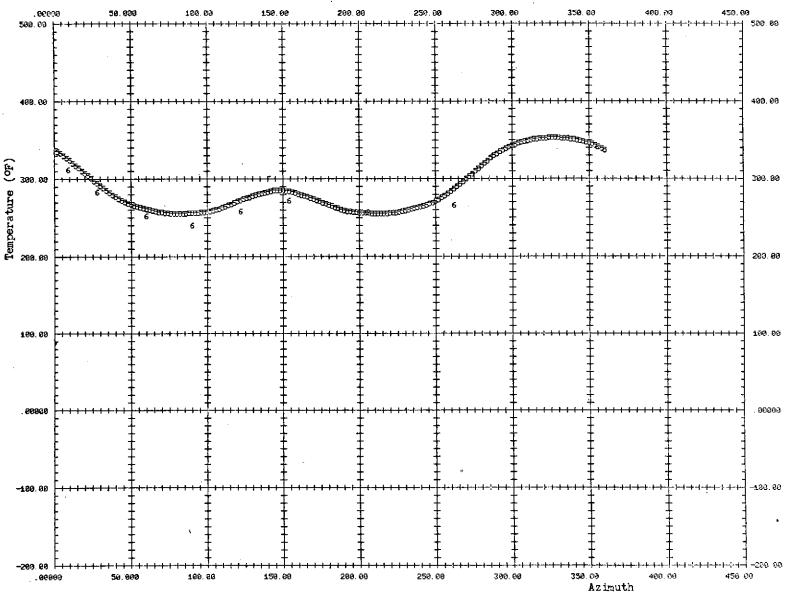


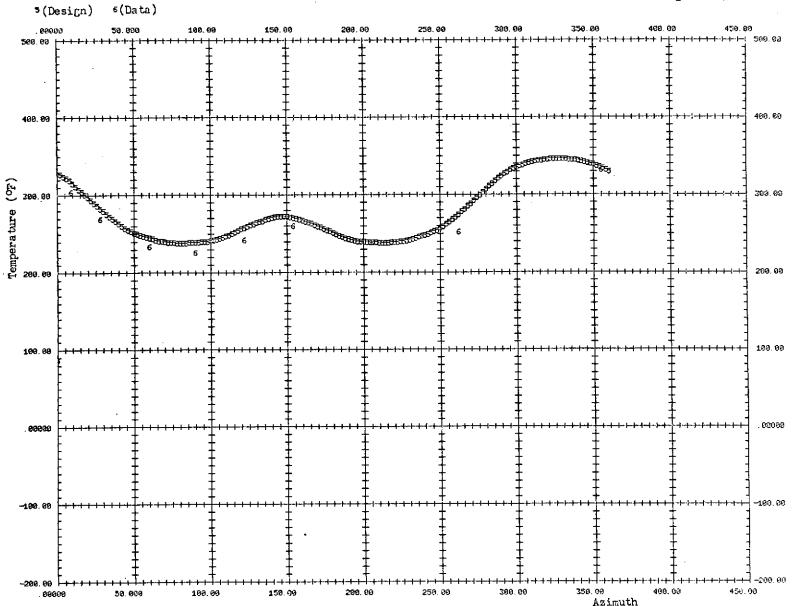
Figure 10(k)

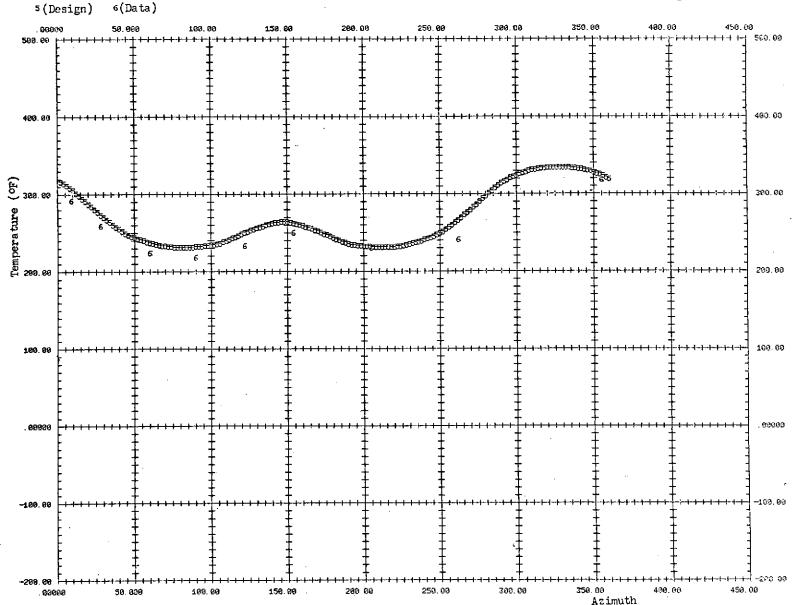


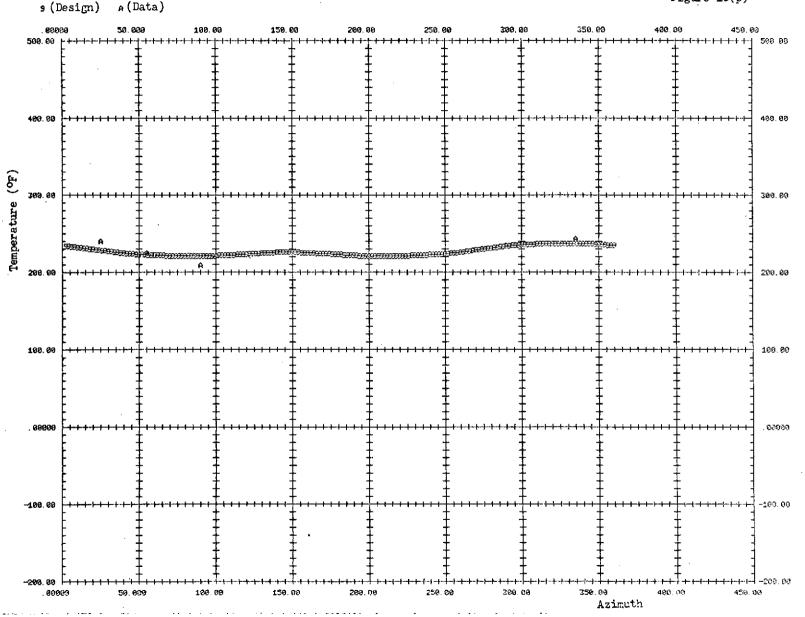






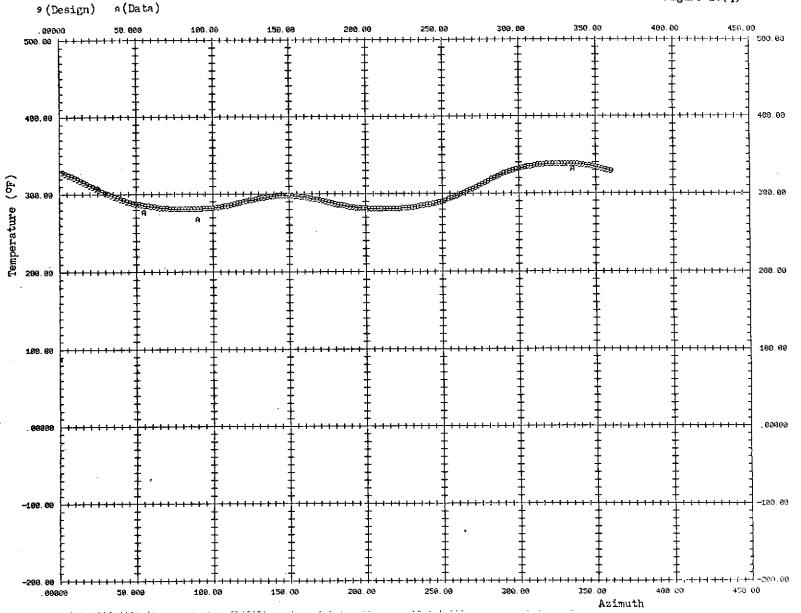






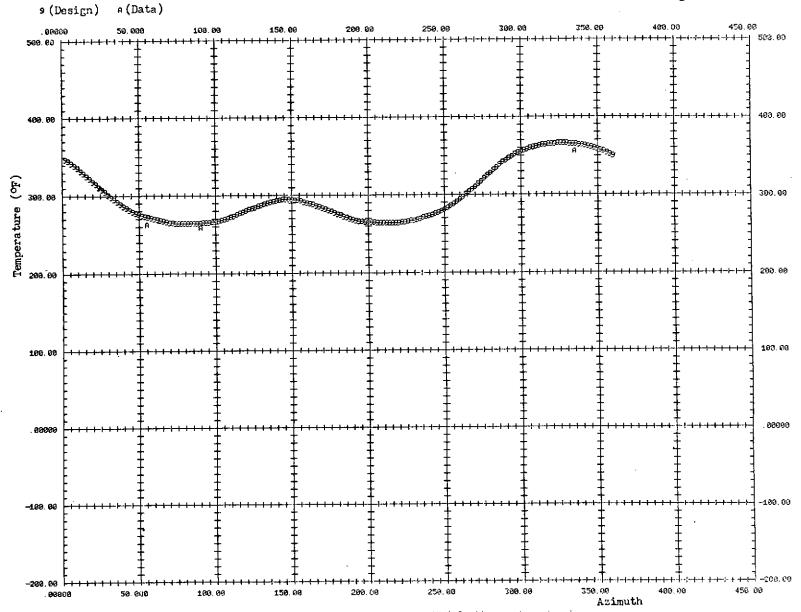
SPF CSS-FST, RUN NO. 42 - CIRC. DIST. PLOT PLOT NUMBER 10 STA. NO. 2552.0, TIME 150.0 TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 10(q)



SPF CSS-FST, RUN NO. 42 - CIRC. DIST. PLOT PLOT NUMBER 10 STA. NO. 2552.0, TIME 200.0 TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 10(r)



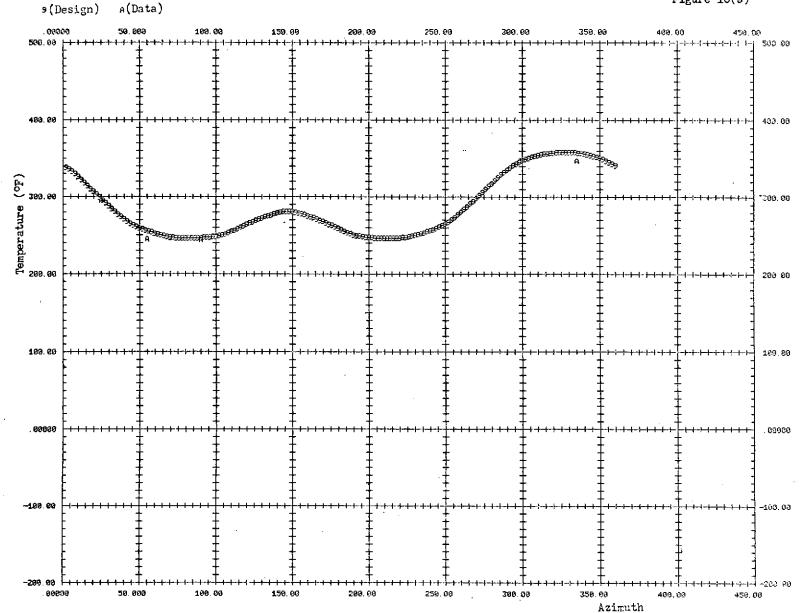
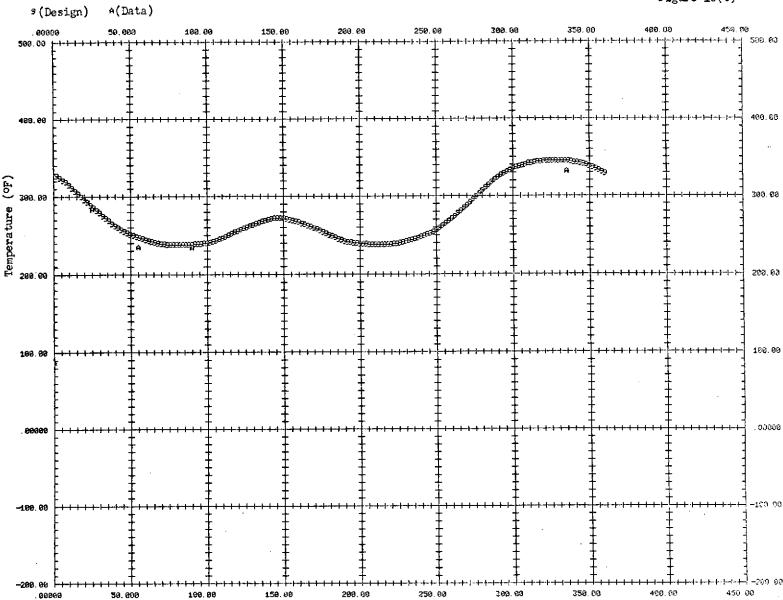


Figure 10(t)

Azimuth



SFF CSS-FST, RUN NO. 42 - CIRC. DIST, PLOT PLOT NUMBER 12 STA. NO. 2626.5, TIME 100.0 TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 10(u)

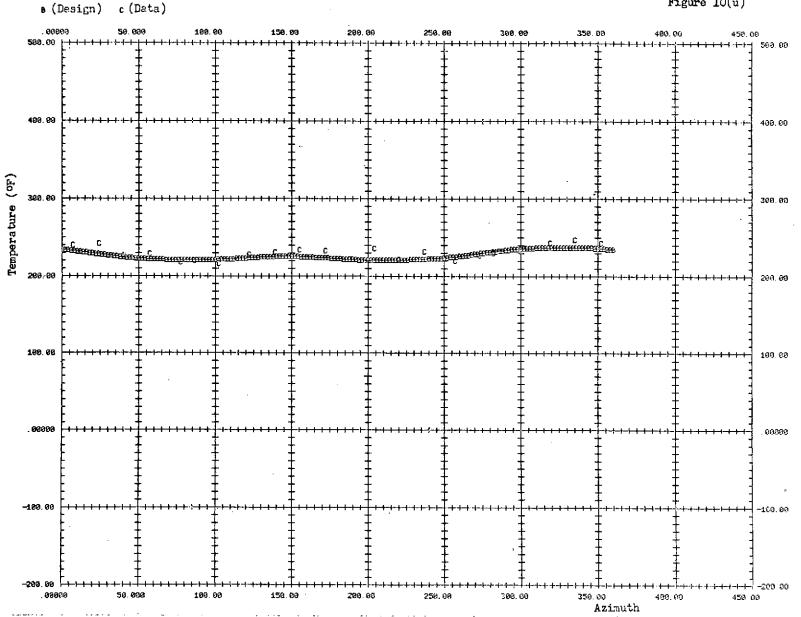
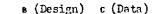
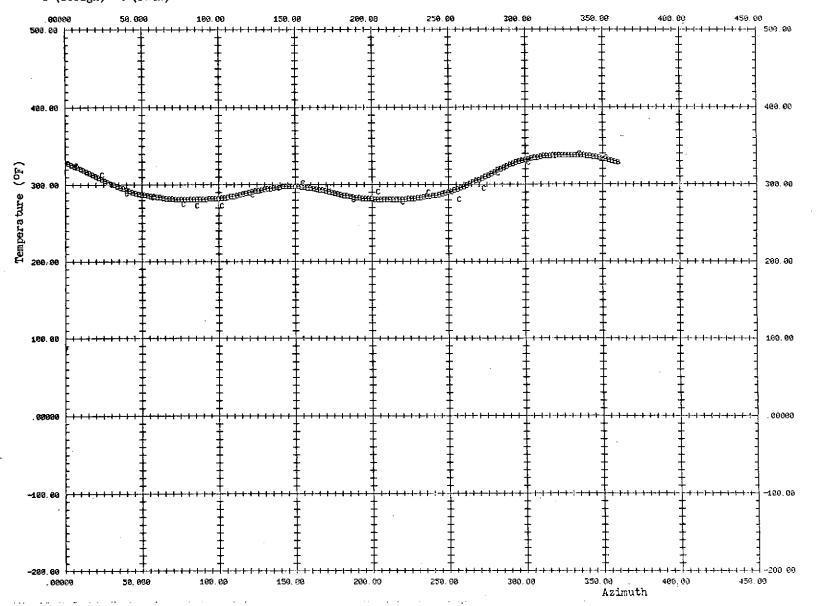


Figure 10(v)





SPF CSS-FST, RUN NO. 42 - CIRC. DIST. PLOT PLOT NUMBER 12 STA. NO. 2626.5, TIME 200.0

TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

Figure 10(w)



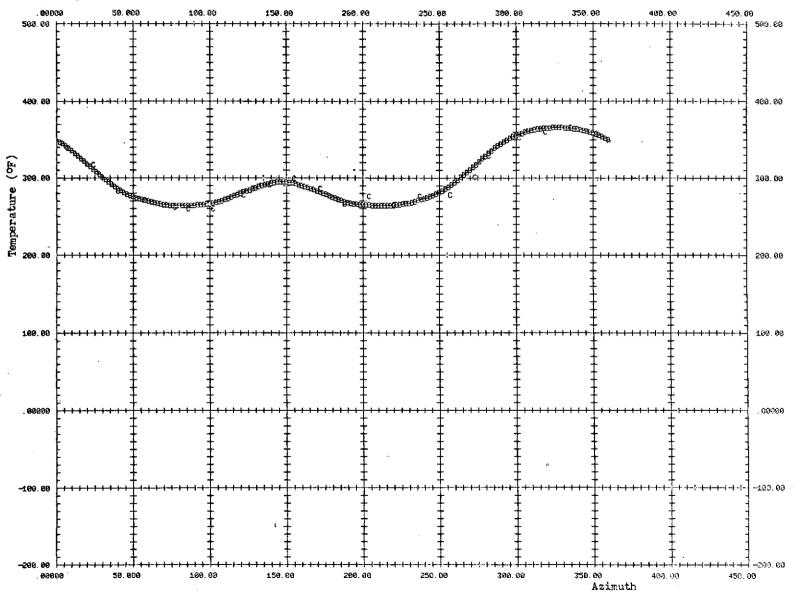
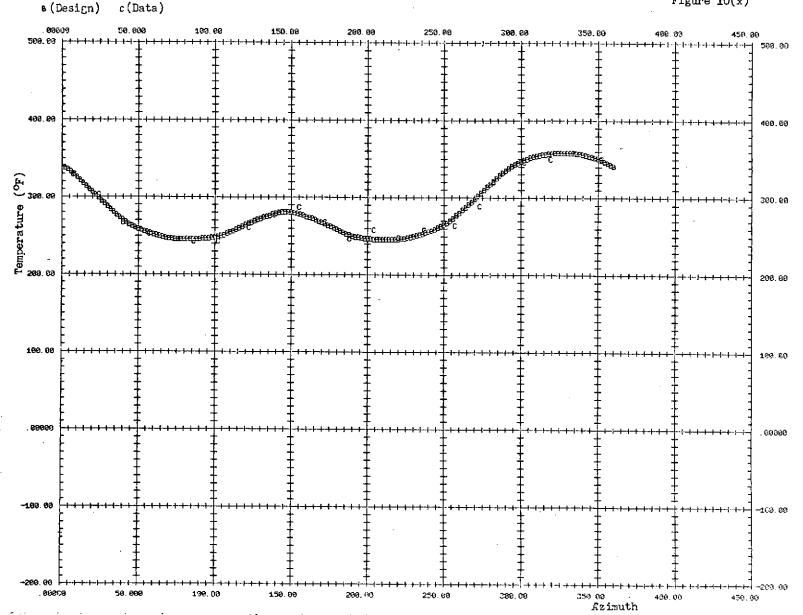
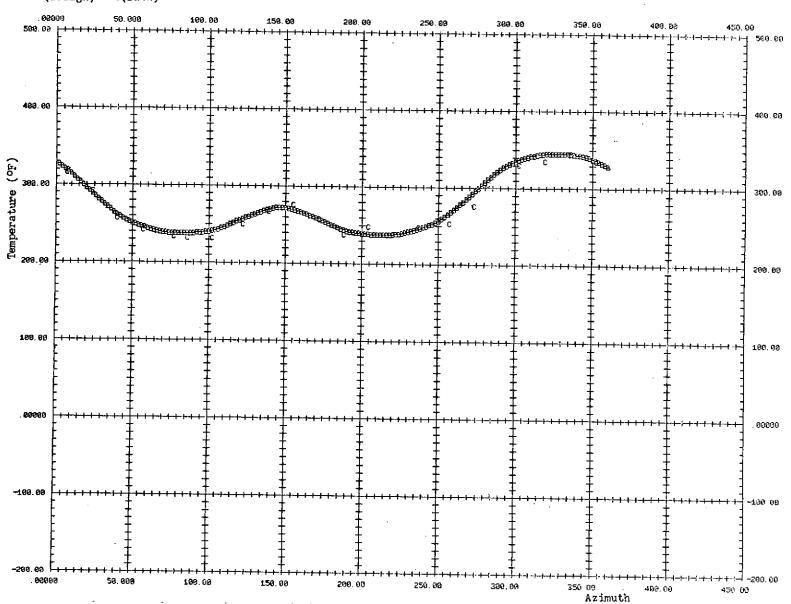
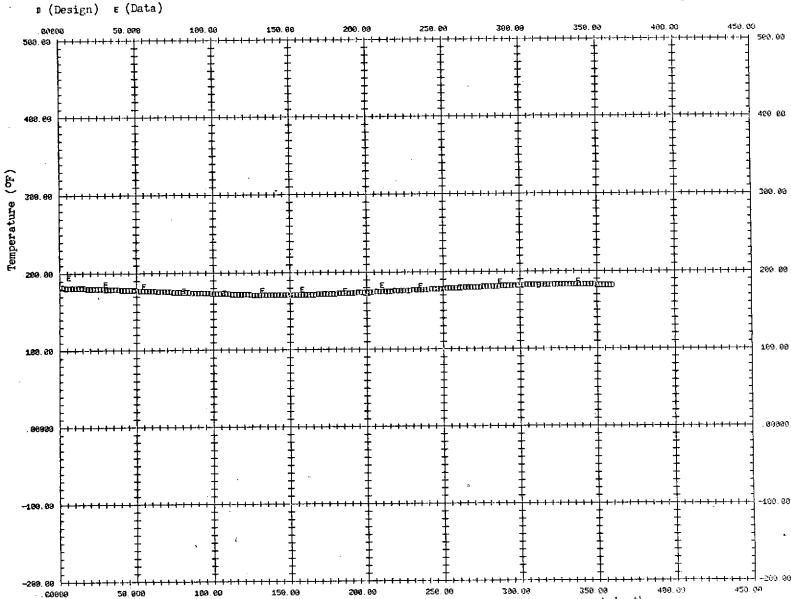


Figure 10(x)

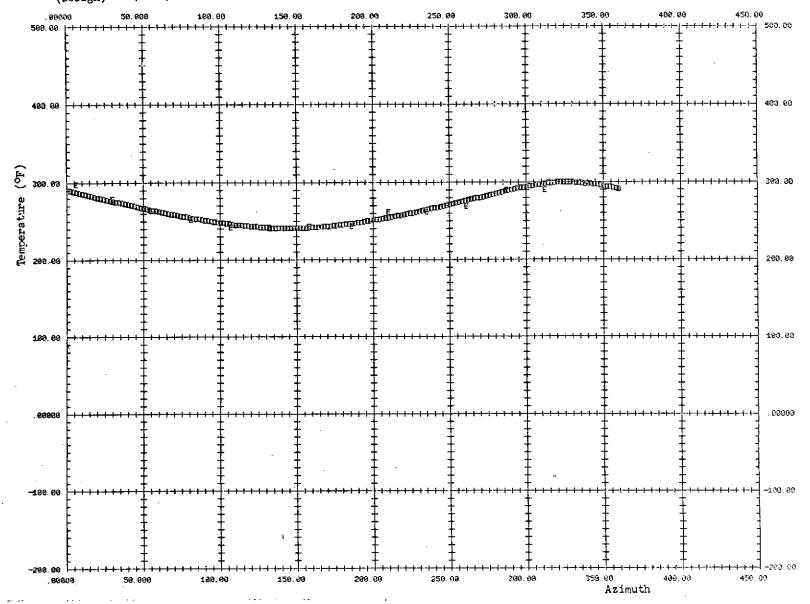




Azimuth



Design) E (Data)



-229 BD 🗀

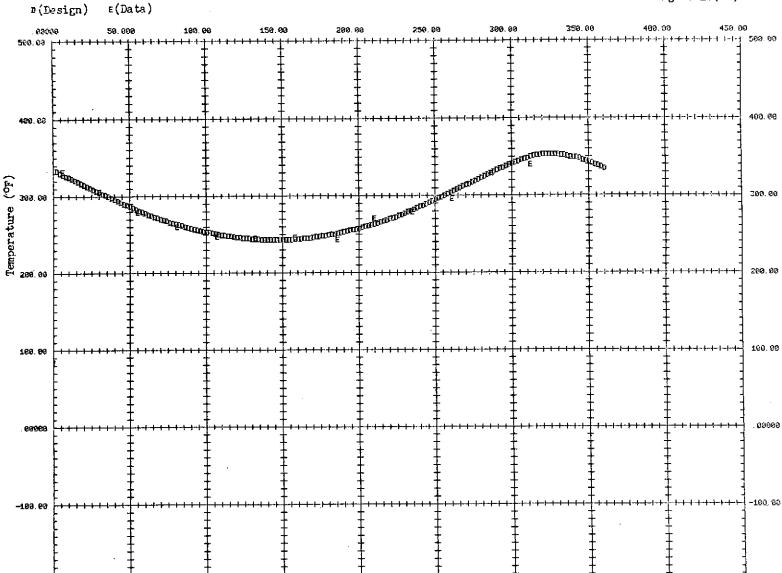
62889

50.960

450 00

400.00

Azimuth



250.00

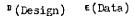
150.00

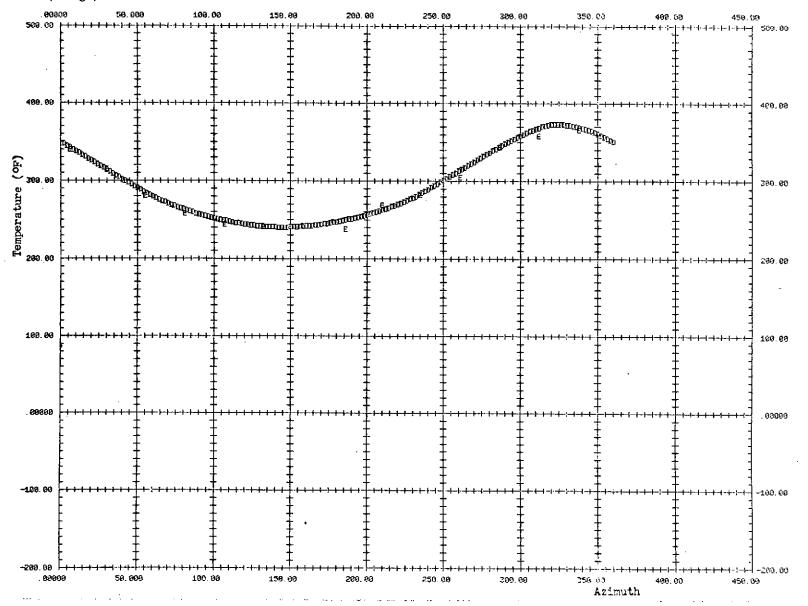
180.09

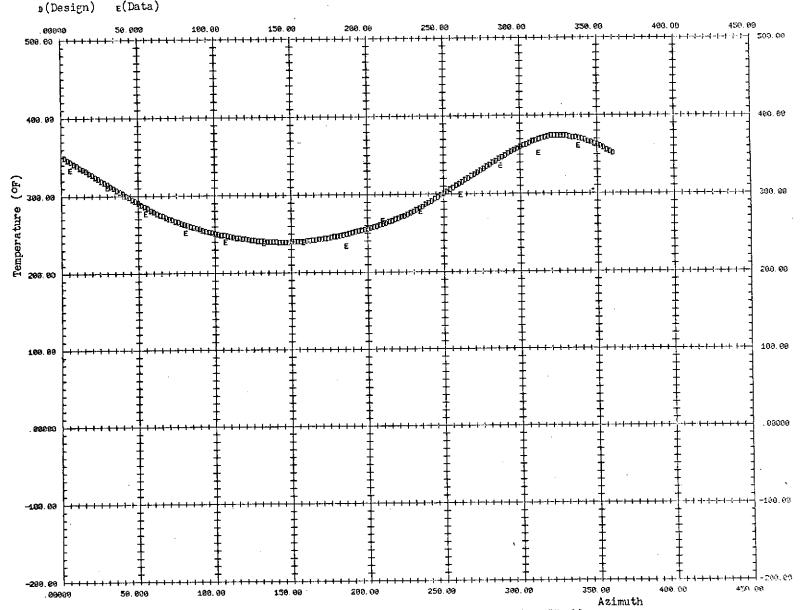
200, 00

300.00

Figure 10(cc)



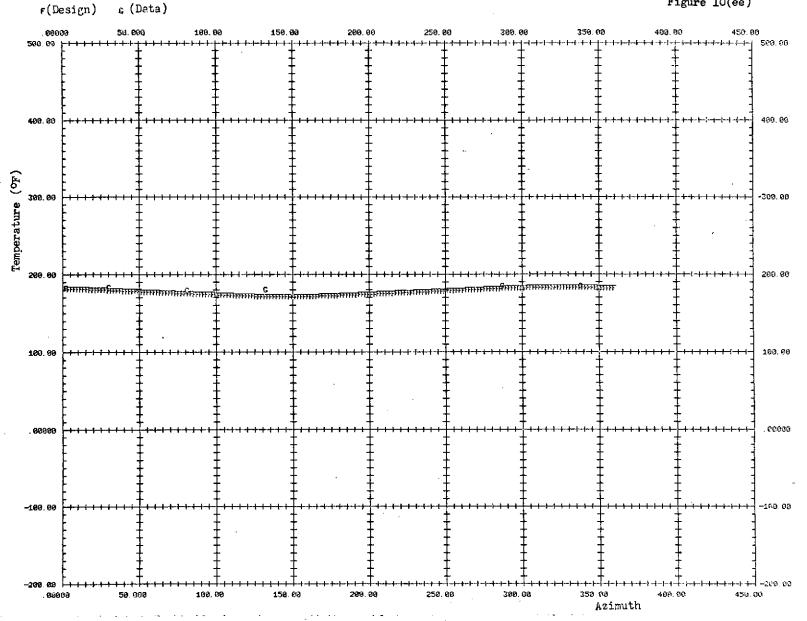


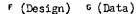


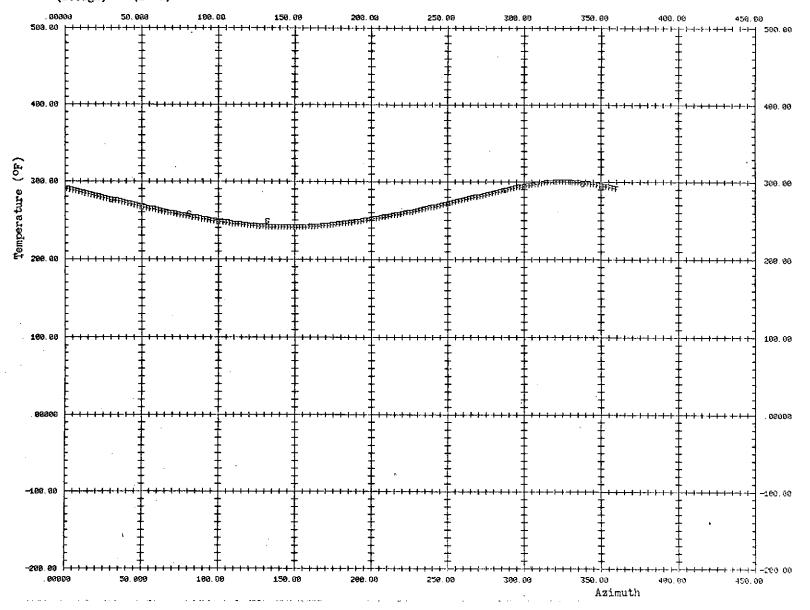
SPF CSS-FST, RUN NO. 42 - CIRC. DIST. PLOT PLOT NUMBER 16 STA. NO. 2792.0, TIME 100.0 FST. PT.323 19 57 11 348

TIME DAY HR MIN SEC MILL

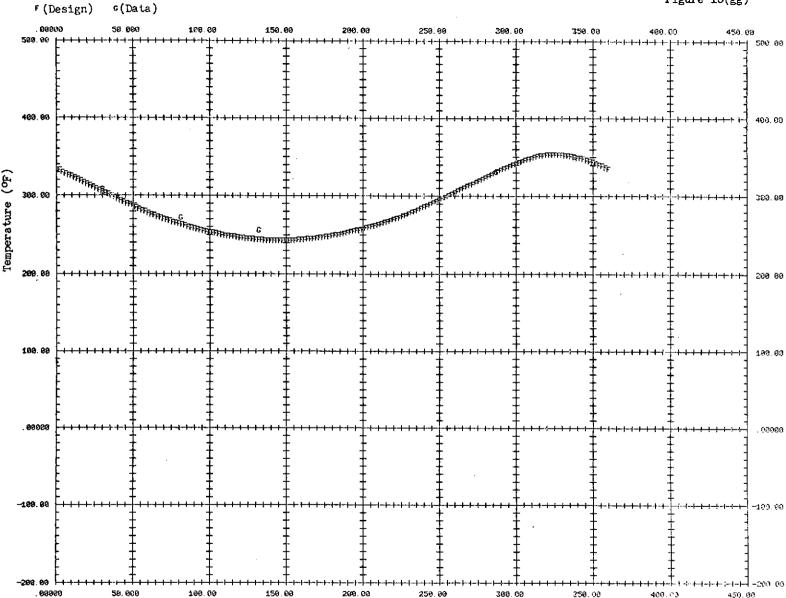
Figure 10(ee)



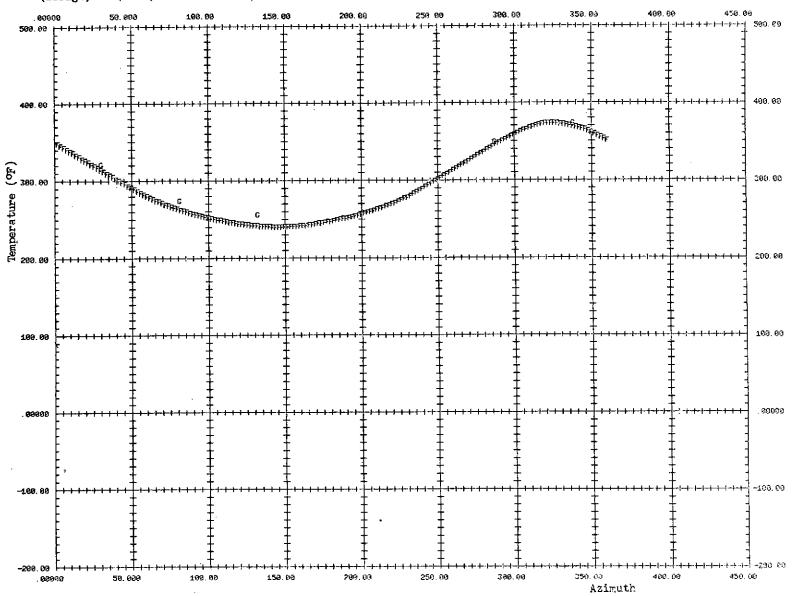




Azimuth



F(Design) C(Data)



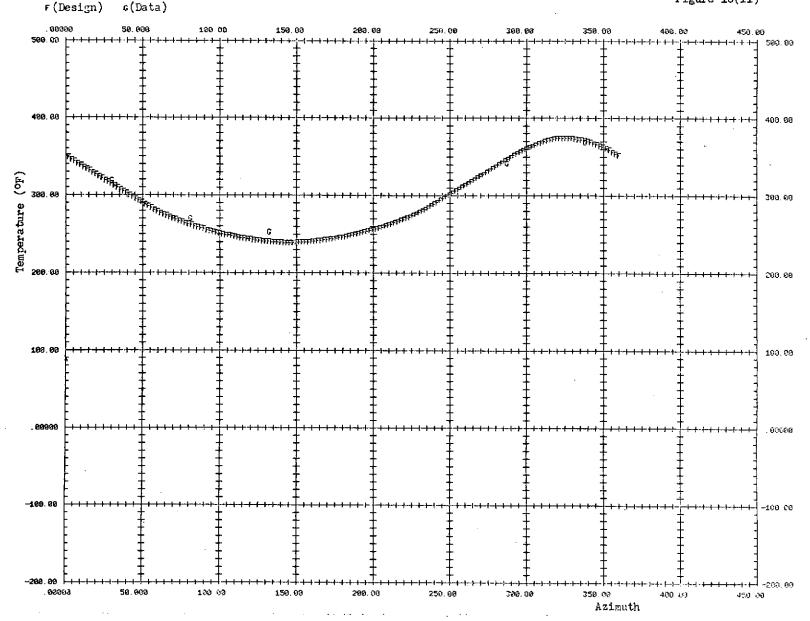
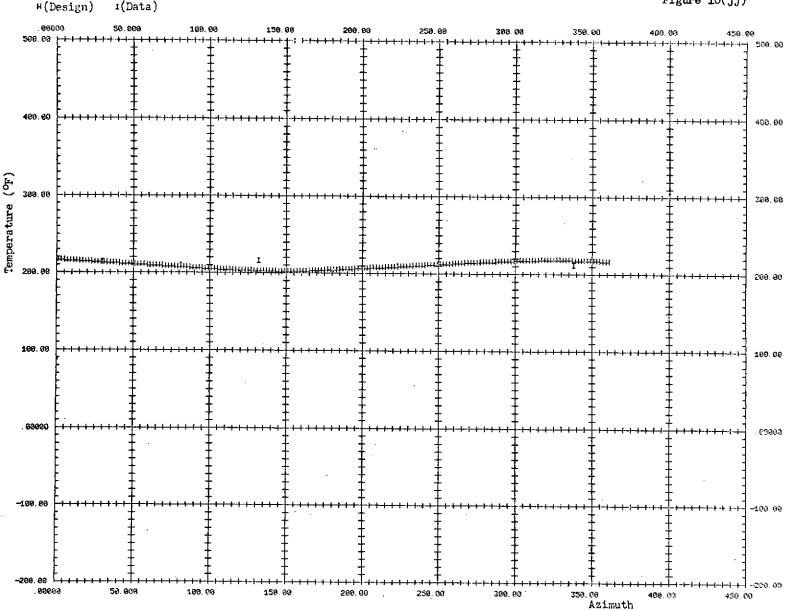
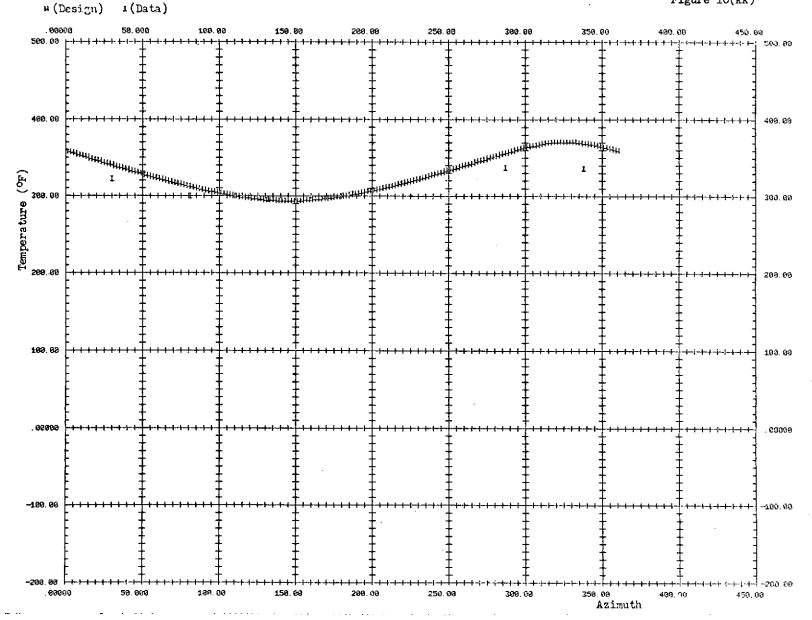
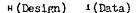


Figure 10(jj)



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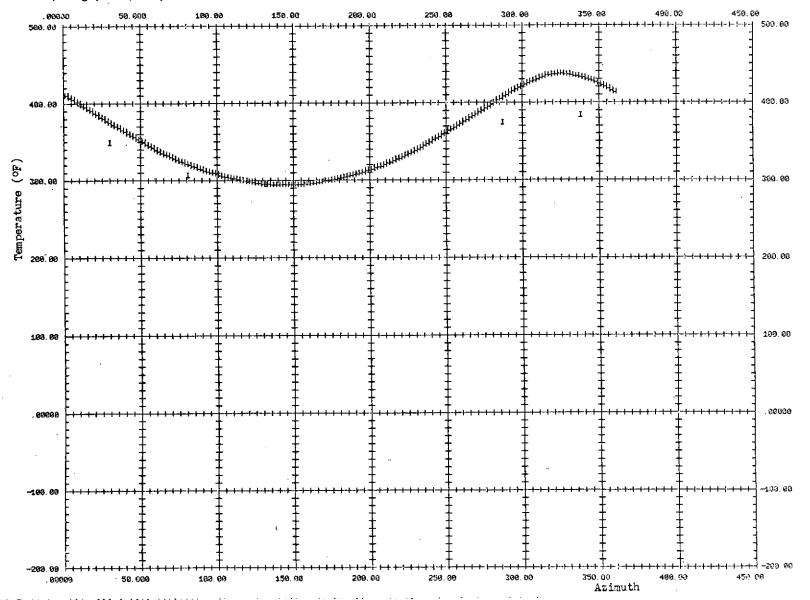


Figure 10(mm)

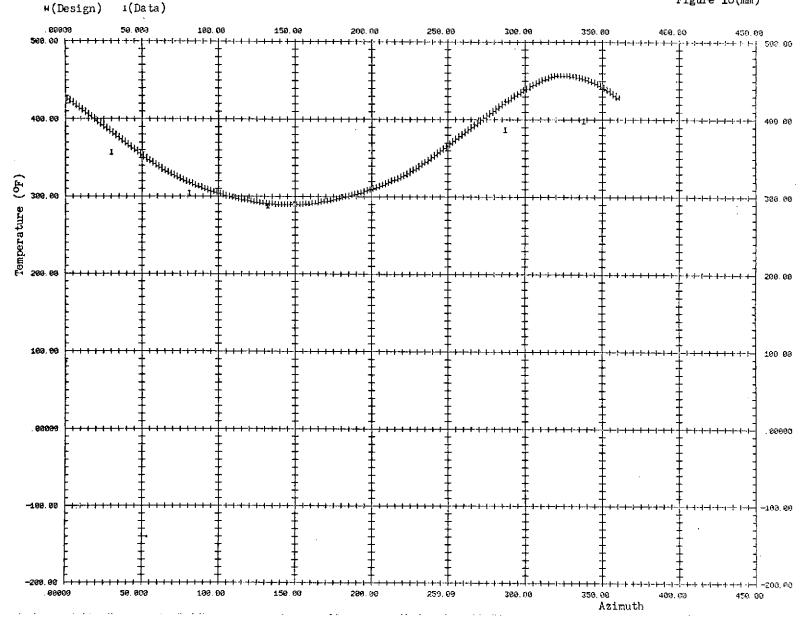
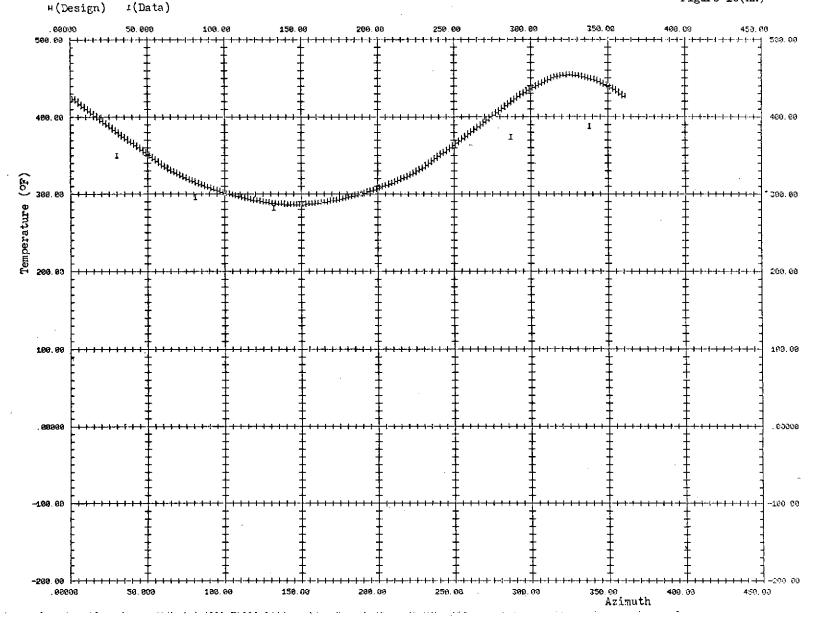
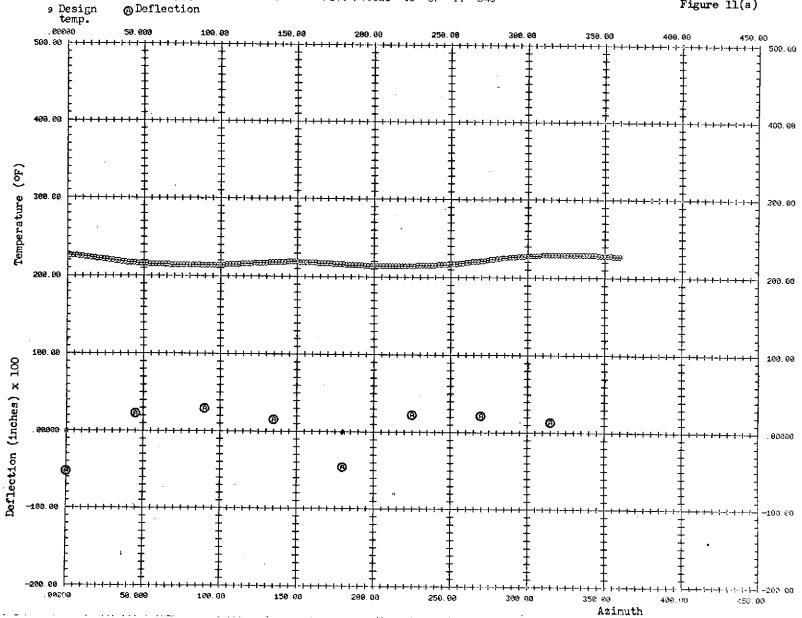
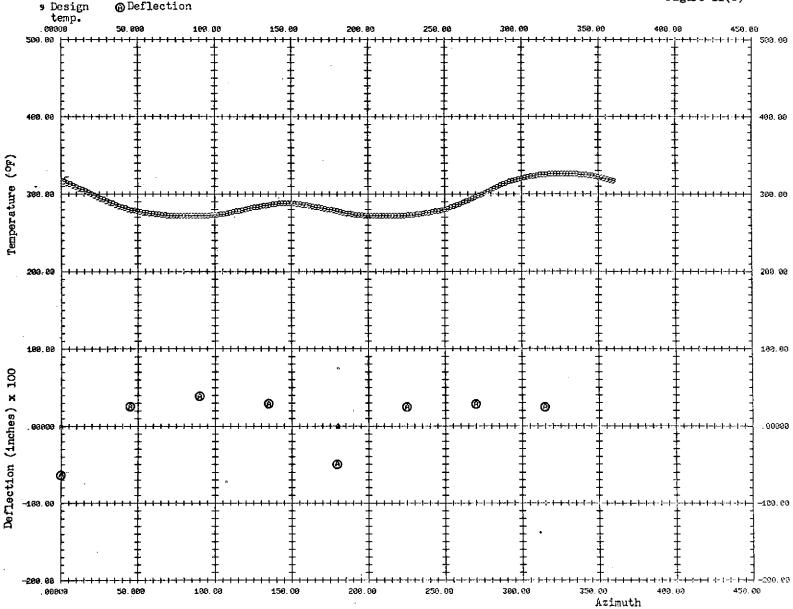


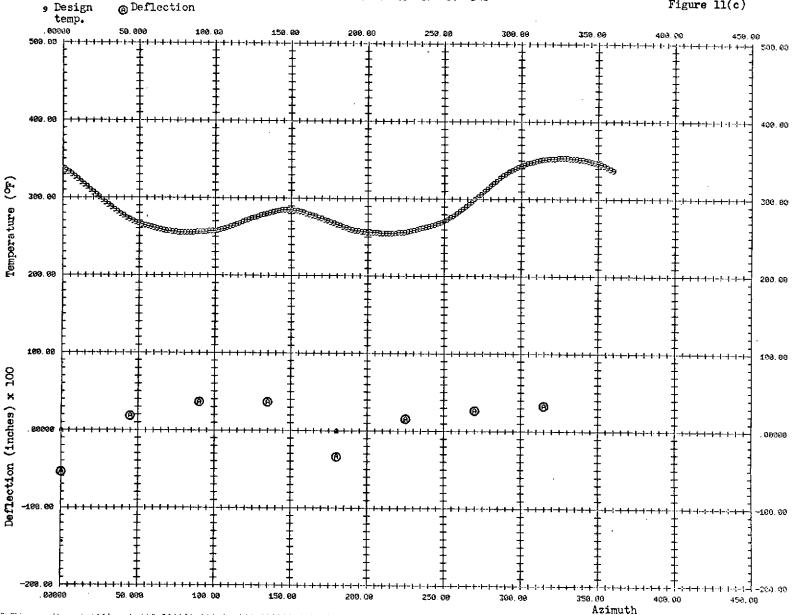
Figure 10(nn)

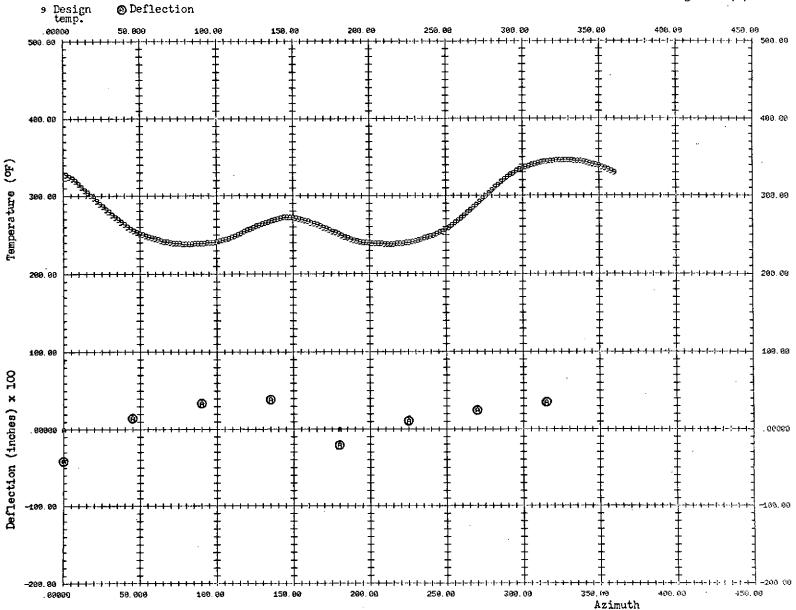


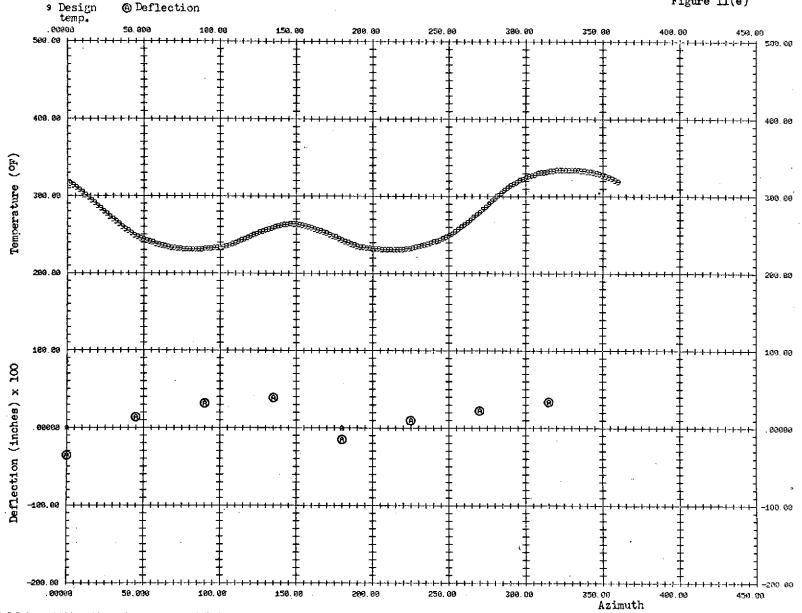
Figures 11(a) thru 11(y). Circumferential deflection and design temperature distributions.

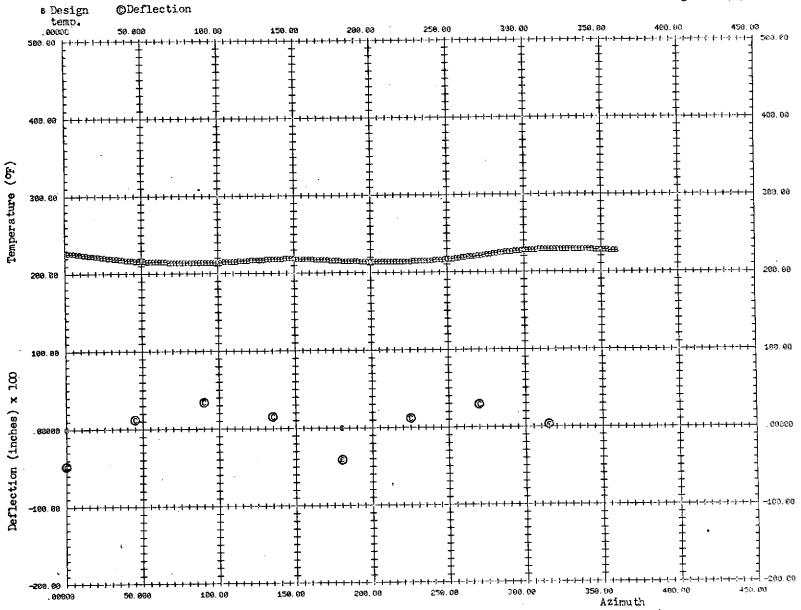


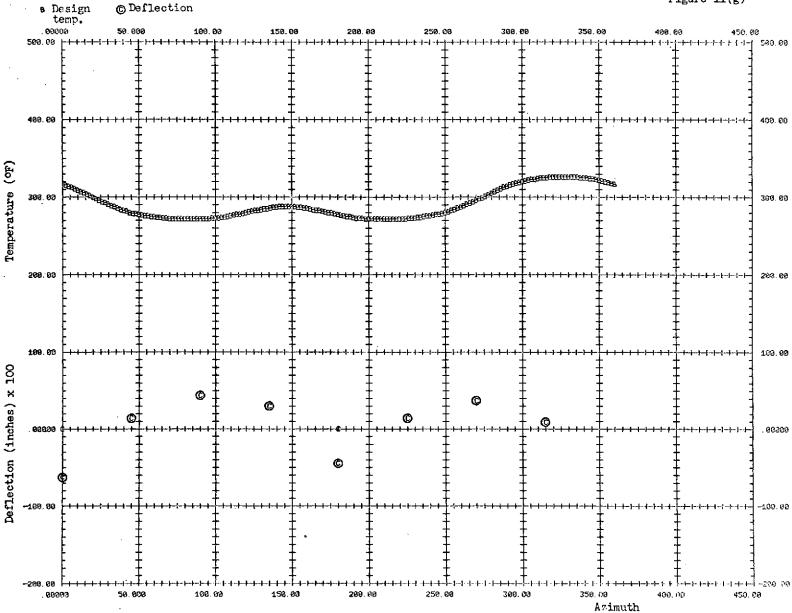












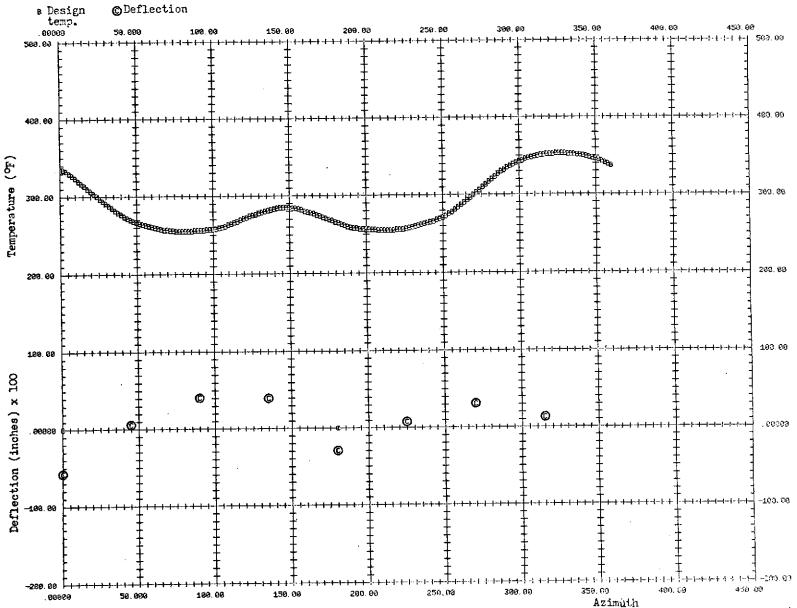
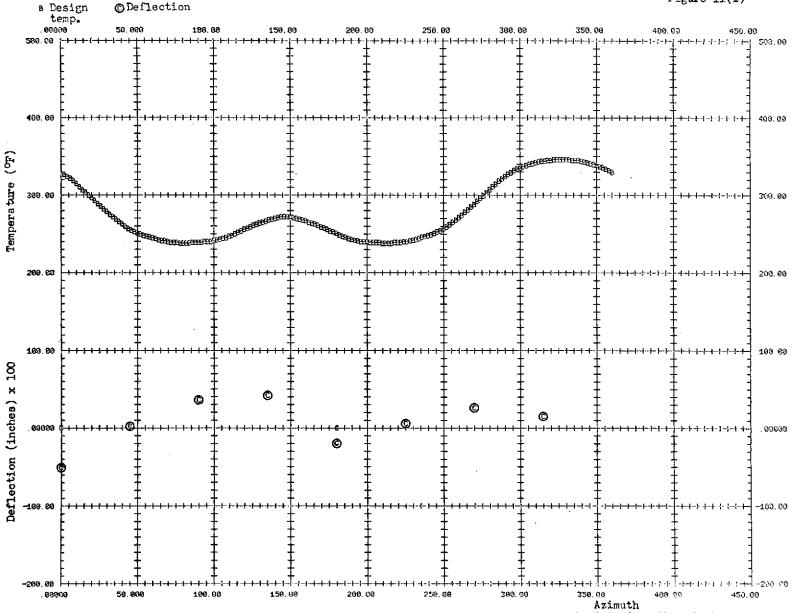
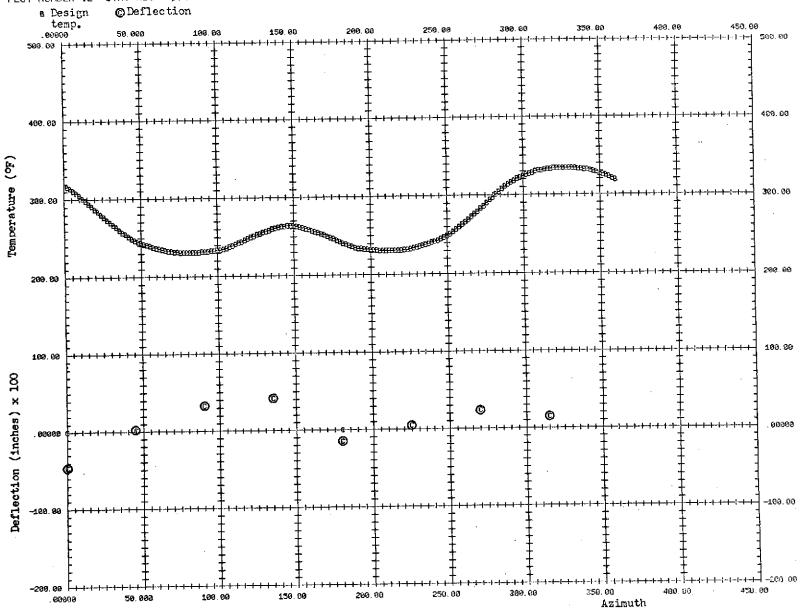
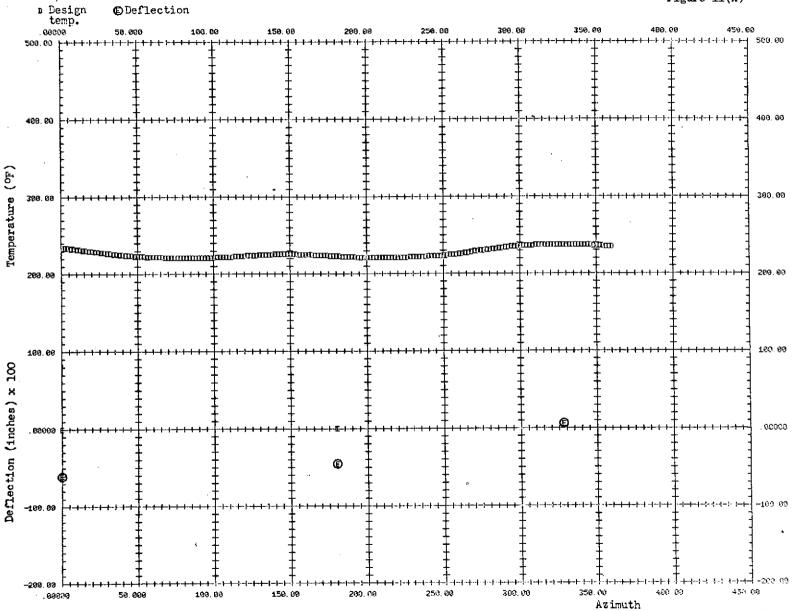
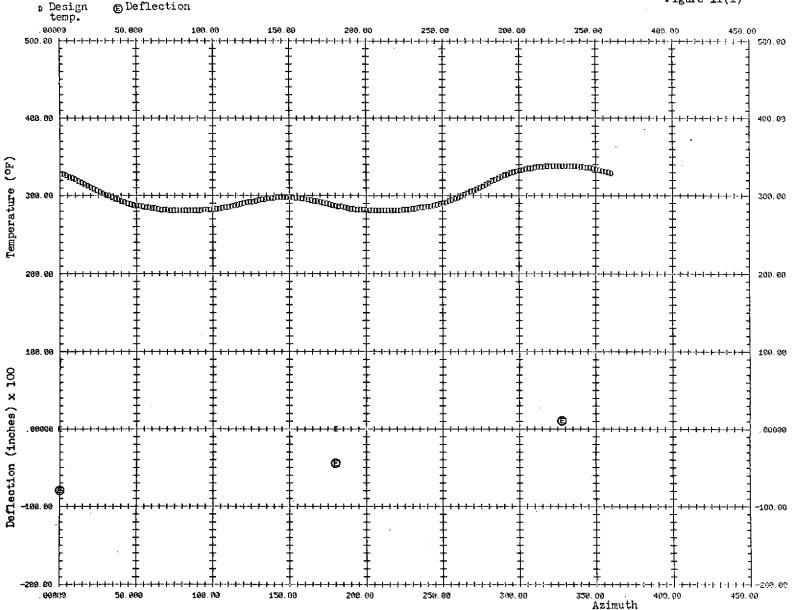


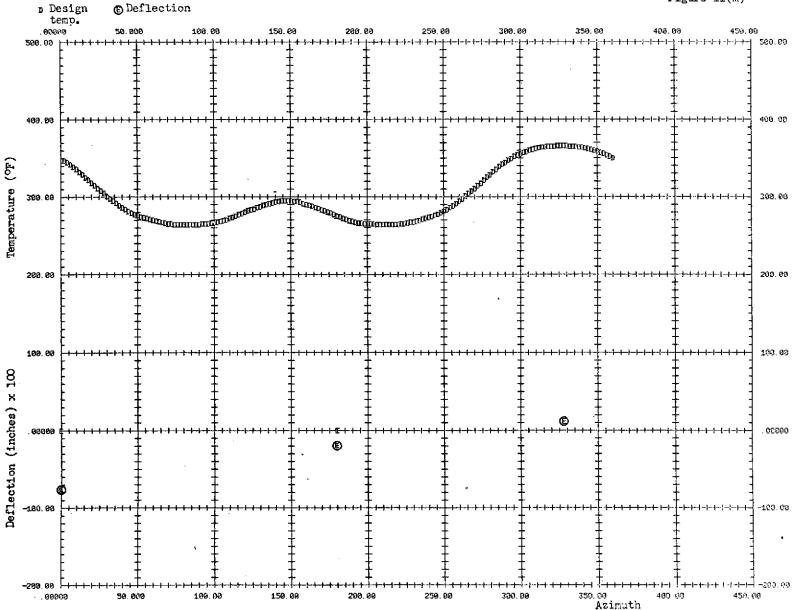
Figure 11(i)

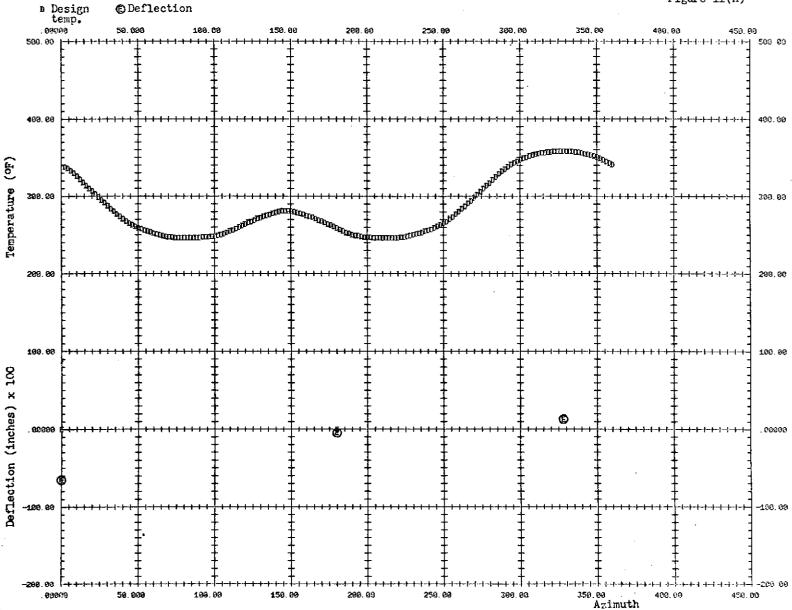


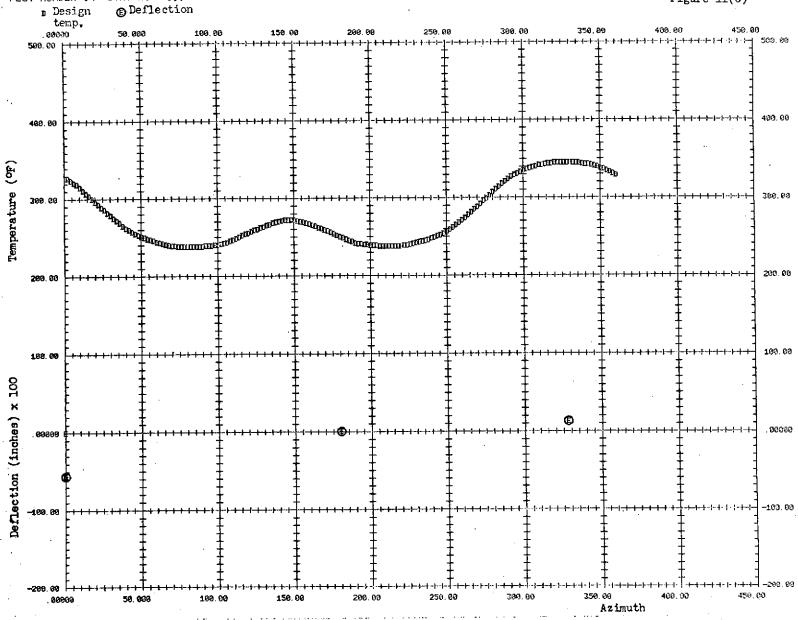








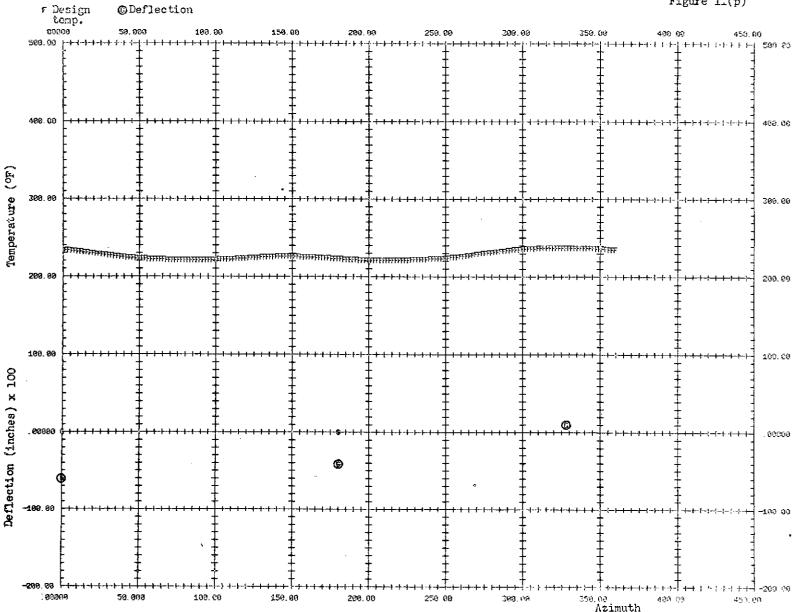




SPE CSS-HST, RUN NO. 42 - CIRC, DIST, PLOT PLOT NUMBER 16 STA. NO. 2619.0, TIME 100.0

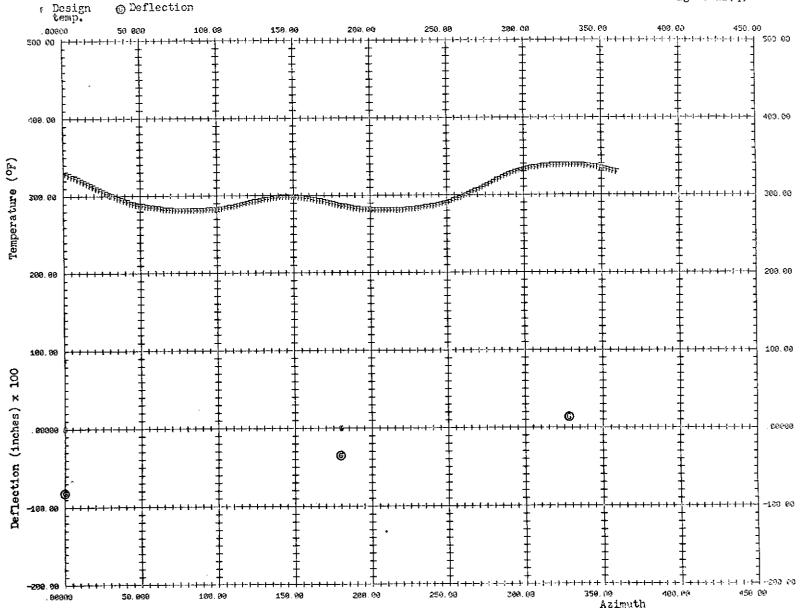
TIME DAY HR MIN SEC MILL. FST. PT.323 19 57 11 348

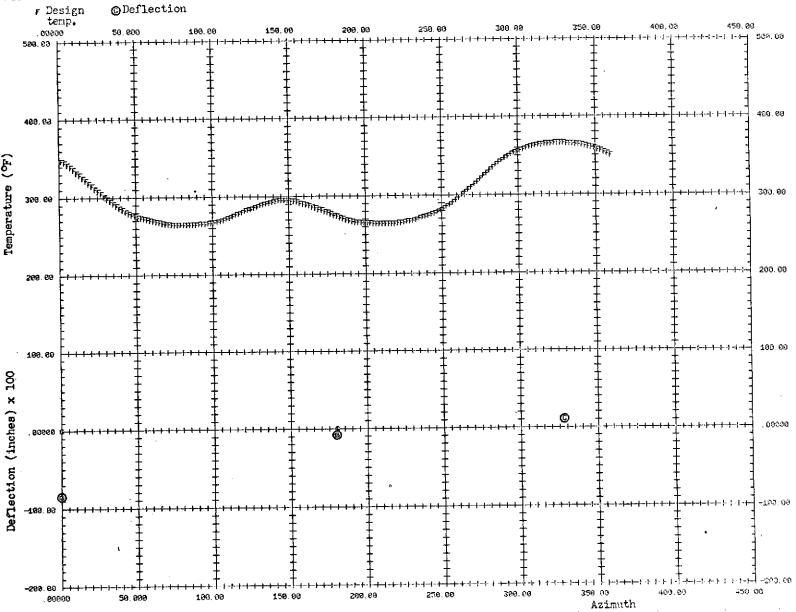
Figure 11(p)

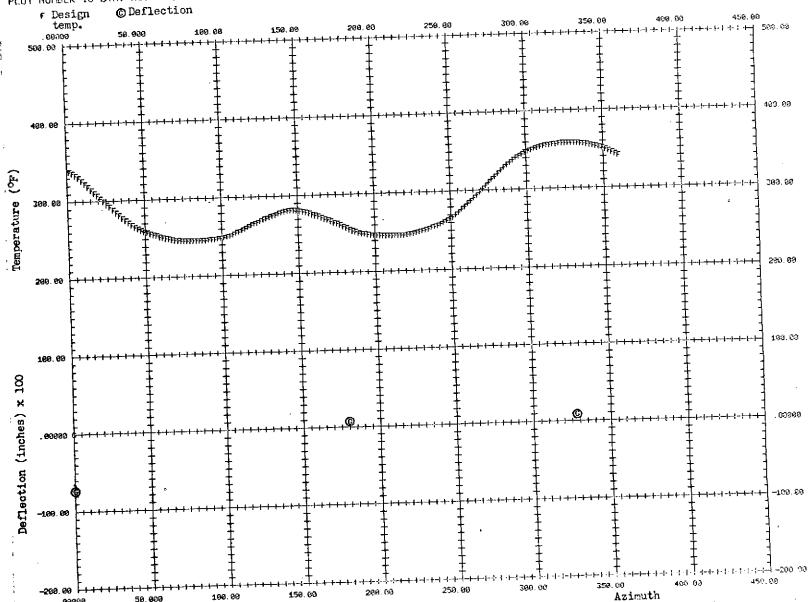


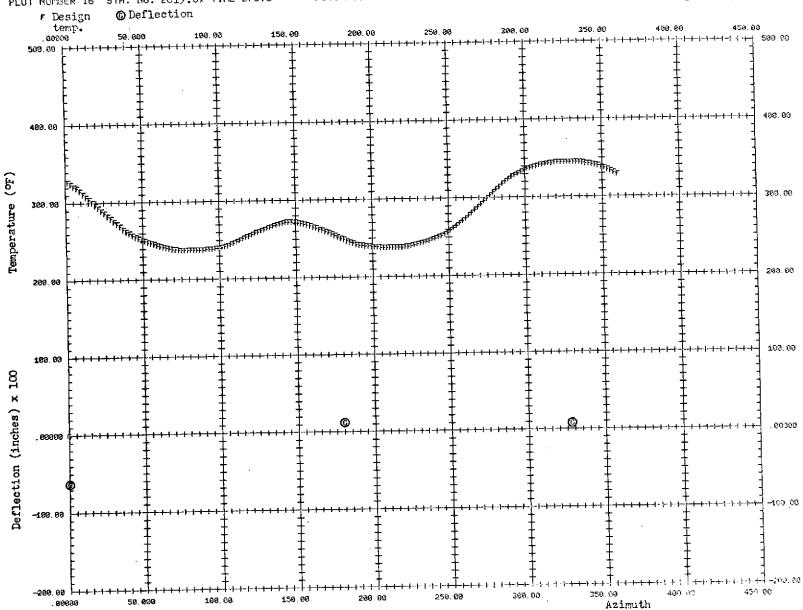
SEE CSS-FST, RUN NO. 42 - CIRC. DIST. PLOT PLOT NUMBER 16 STA. NO. 2619.0, TIME 150.0 TIME DAY HR MIN SEC MILL FST. PT.323 19 57 11 348

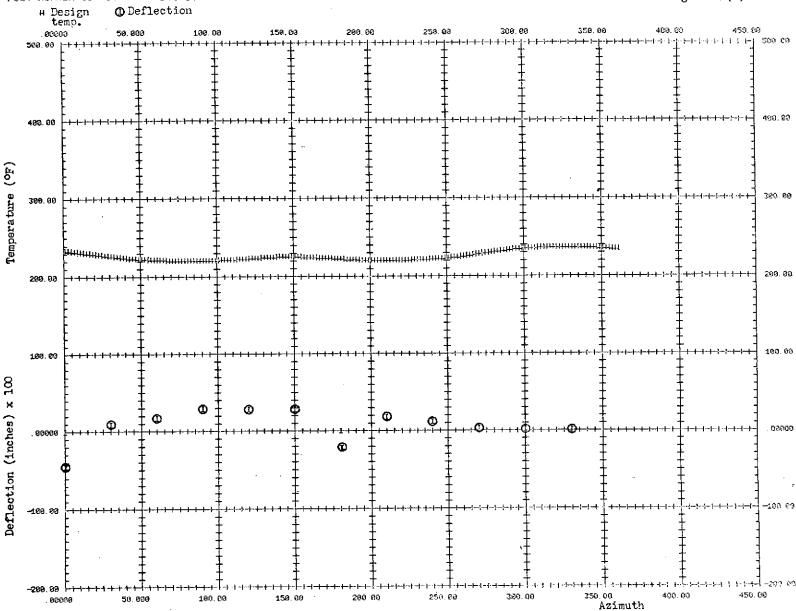
Figure 11(q)

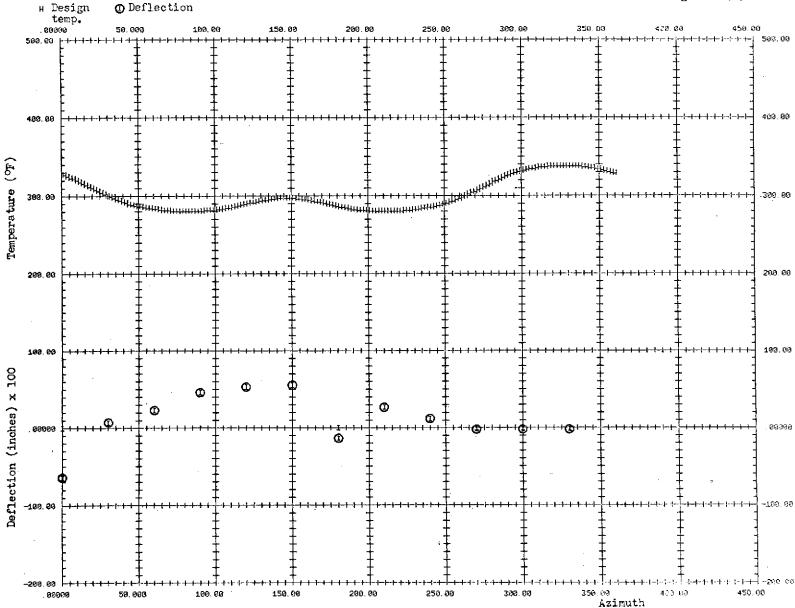












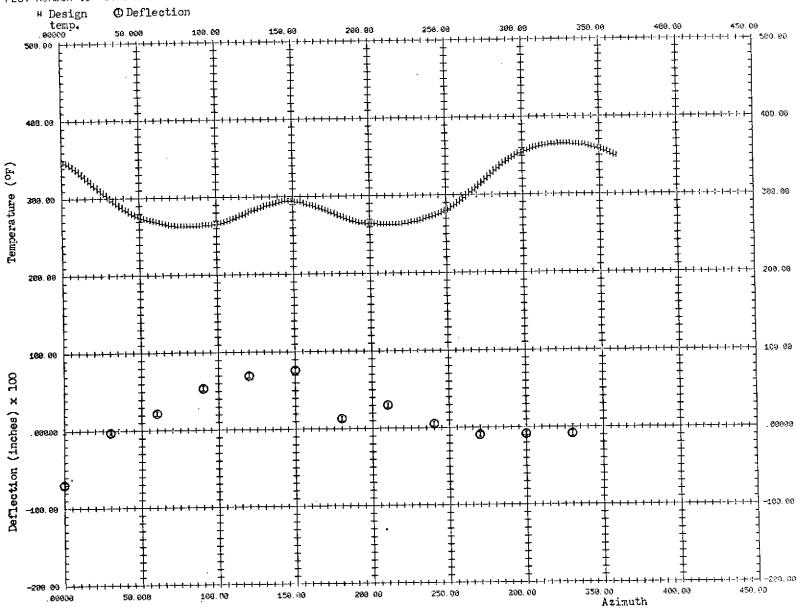


Figure 11(x)

